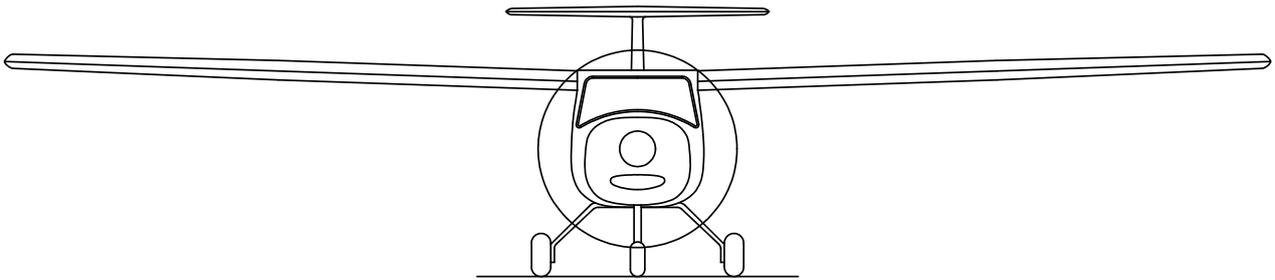




PIPISTREL

Pilot's Operating Handbook

**applies to ALPHA Trainer - registered as UL,
equipped with Rotax 912 (80 HP) engine**



REVISION A06
(16 th August 2023)

SERIAL NUMBER _____

REGISTRATION NUMBER _____

For additional document applicability information, please refer to:
SB-160-00-80-990 Status of continuing airworthiness documentation

WARNING!

This document **MUST** be present inside the cockpit at all times.
Should you sell this aircraft make sure this document is given to the new owner.

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Performance - specifications

ALPHA Trainer	80 hp Rotax 912
Stall speed (flaps extended)	34 kts (64 km/h)
Stall speed (flaps retracted)	40 kts (74 km/h)
Cruise speed (5300 RPM)	108 kts (201 km/h)
Maximum horizontal speed at sea level	120 kts (222 km/h)
VNE	135 kts (250 km/h)
Standard endurance 108 knots cruise	3.0 hours + 30 min reserve
Standard range at 108 knots cruise	324 NM (600 km)
Fuel flow at 108 knots cruise speed (typical)	3.6 gph (13.6 l/h)
Range at best range cruise speed (90 KIAS 4900-5000 RPM)	380 NM (702 km)
Takeoff - ground roll - at MTOM	459 ft (140 m)
Takeoff total distance over 50 ft obst. at MTOM	738 ft (225 m)
Landing distance over 50 ft obst.	1510 ft (460 m)
Absolute ceiling at MTOM	18,000 ft (5500 m)

NOTE The above performance figures are based on an airplane weight at 1040 lbs, standard atmospheric conditions, level hard-surfaced dry runways and no wind. They are calculated values derived from flight test conducted by Pipistrel d.o.o. , under supervision of the Slovenian CAA and under carefully documented conditions. Figures may vary based on numerous factors (surface condition, temperature, water on wing, etc.).

ALPHA Trainer	80 hp Rotax 912
Maximum weight takeoff	1040 lbs (472.5 kg)
Maximum weight landing	1040 lbs (472.5 kg)
Typical empty weight	684 lbs (310 kg)
Baggage allowance, maximum (baggage area floor limit)	22 lbs (10 kg)
Fuel capacity, total	13.2 USgal (50 l)
Fuel capacity, usable	12.7 USgal (48 l)
Oil capacity (oil bottle)	3 liters
Engine	Rotax 912 80 hp
Propeller	Pipistrel FP02-80, dia. 65" 1660 mm

Coverage

The Pilot's Operating Handbook (POH), found in the airplane at the time of delivery from Pipistrel d.o.o. contains information applicable to the ALPHA Trainer airplane and to the airframe designated by the serial number and registration number shown on the title page. All information is based on data available at the time of publication.

This POH consists of ten sections that cover all operational aspects of a standardly equipped airplane. Section 10 contains the supplements which provide amended operating procedures, performance data and other necessary information for those airplanes that conduct special operations and/or are equipped with both standard and optional equipment. Additional supplements are individual documents which are issued/revised separately. The log of effective pages should be used to determine the status of each supplement.

Revision tracking, filing and identifying

Pages to be removed or replaced in the Pilot's Operating Handbook are determined by the log of effective pages located in this section. This log contains the page number and revision number for each page within the POH. As revisions to the POH occurs, the revision number on the effected pages is updated. When two pages display the same page number, the page with the latest revision shall be used in the POH. The revision number on the log of effective pages shall also coincide with the revision number of the page in question. As an alternative to removing and/or replacing individual pages, the owner can also print out a whole new manual in its current form, which is always available from www.pipistrel-aircraft.com.

Revised material is marked with a vertical double-bar that will extend the full length of deleted, new, or revised text added to new or previously existing pages. This marker will be located adjacent to the applicable text in the marking on the outer side of the page. The same system applies when the header, figure, or any other element inside this POH is revised. A list of revisions is located at the beginning of the log of effective pages. Pipistrel is not responsible for technical changes/updates to OEM manuals supplied with the aircraft (eg. radio, transponder, GPS, etc.).

Online updates, service notice tracking & airworthiness reporting

To receive and report Issues and anomalies identified by the aircraft owner or maintainer during the operation or maintenance of this aircraft or to report content errors in this manual then please log into the Owner's section of the Pipistrel website where you can report service difficulties and receive updates and information relevant to service and airworthiness of your aircraft.

Go to: www.pipistrel-aircraft.com and log in the top right corner of the page with:

Username: owner1

Password: ab2008

Index of revisions

The table below shows the revision history of this POH. Check your registration authority's or the manufacturer's website occasionally for the most up-to-date releases of operation-relevant documentation, which includes this POH.

Revision	Reason for revision	Affected pages	Issuer, release date
REV 0.	/	/	Tomazic, Pipistrel d.o.o. Ajdovščina, 15th Nov 2011
REV 1.	Reorganisation of chapters, content adapted for LTF-UL2003 compliance and Austrocontrol national requirements	ALL	Tomazic, Pipistrel d.o.o. Ajdovščina, 29th May 2015
REV 2.	Placards, Weight and Balance, Typical Instrument Panel, Electric airbrakes added, Preflight check amended, procedure amended	See log of effective pages	SLO.DOA.002, 5th March 2021
A03	Adding fuel scheme, editorial amendments	ALL	SLO.DOA.002, 25th Nov 2021
A04	Fuel and oil specifications, brake fluid, manoeuvre limits, minimum equipment list, baggage weight limit, instrument markings, stall & spin recovery amended, best glide ratio flaps position, ELT check, refuelling, IAS-CAS calibration, CG range, speed corrections	Cover, i-1 to 3 to i-7, 0-2, 1-2, 1-4 to 5, 2-3 to 6, 2-9, 3-2 to 4, 4-5 to 10, 5-2, 5-5, 6-3, 6-5 to 6, 7-2 to 3, 7-5 to 6, 7-8 to 9, 7-11, 12-2, 3-6, 6-3, 6-6, 8-2, 9-3, 9-14 to back cover	SLO.DOA.002, 26th Sept 2022
A05	Placards, kinds of operations, refuelling procedure	Cover, i-1, i-4 to 6, 2-6, 2-8 to 11, 4-7, 8-2 to 3	SLO.DOA.002, 2nd March 2023
A06	VNO speed designation corrected as VB, Preflight cables check, typical empty weight, ELT system description, switch labels updated, airplane description amended, checklist improved, editorial corrections	Cover, i-2, i-4 to 6, 1-4, 1-5, 2-2, 3-3, 3-4, 4-3, 4-5, 4-6 to 4-10, 7-1, 7-4 to 7-6, 7-9, 7-11 to 14, 8-4, 9-2, 9-5, 9-15.	SLO.DOA.002, 16th Aug 2023

Warnings, Cautions and Notes

Safety definitions used in the manual:

WARNING! An operating procedure or technique that may result in personal injury or loss of life if not followed.

CAUTION! An operating procedure or technique that may result in damage to equipment if not followed.

NOTE An operating procedure or technique needing special emphasis.

Log of effective pages

Use the table below to determine the currency and applicability of your POH. Pages affected by the current revision are marked in bold text in the page number column.

Page number	Page status	Rev. number	Page number	Page status	Rev. number
Cover	Revised	-	4-6	Revised	5
i-1	Blank Page	-	4-7	Revised	6
i-2	Revised	5	4-8	Revised	5
i-3	Revised	4	4-9	Revised	5
i-4	Revised	5	4-10	Revised	5
i-5	Revised	5	5-1	Revised	3
i-6	Revised	5	5-2	Revised	4
i-7	Blank Page	-	5-3	Revised	3
0-1	Revised	3	5-4	Revised	3
0-2	Blank Page	-	5-5	Revised	4
1-1	Revised	3	5-6	Revised	3
1-2	Revised	4	6-1	Revised	3
1-3	Revised	3	6-2	Revised	3
1-4	Revised	5	6-3	Revised	4
1-5	Revised	5	6-4	Revised	3
1-6	Blank Page	-	6-5	Revised	4
2-1	Revised	3	6-6	Blank Page	-
2-2	Revised	5	7-1	Revised	4
2-3	Revised	5	7-2	Revised	4
2-4	Revised	4	7-3	Revised	4
2-5	Revised	4	7-4	Revised	4
2-6	Revised	5	7-5	Revised	5
2-7	Revised	3	7-6	Revised	5
2-8	Revised	4	7-7	Revised	4
2-9	Revised	5	7-8	Revised	5
2-10	Revised	4	7-9	Revised	5
2-11	Revised	4	7-10	Revised	3
2-12	Blank Page	-	7-11	Revised	5
3-1	Revised	3	7-12	Revised	4
3-2	Revised	4	7-13	Added	0
3-3	Revised	5	7-14	Blank page	-
3-4	Revised	5	8-1	Revised	3
3-5	Revised	4	8-2	Revised	6
3-6	Blank Page	-	8-3	Revised	4
4-1	Revised	3	8-4	Revised	4
4-2	Revised	3	8-5	Revised	3
4-3	Revised	4	8-6	Revised	3
4-4	Revised	3	9-1	Revised	3
4-5	Revised	5	9-2	Revised	4

Log of effective pages (continued)

Page number	Page status	Rev. number
9-3	Revised	4
9-4	Revised	3
9-5	Revised	4
9-6	Revised	3
9-7	Revised	3
9-8	Revised	3
9-9	Revised	3
9-10	Revised	3
9-11	Revised	3
9-12	Revised	3
9-13	Revised	3
9-14	Blank Page	-
9-15	Revised	5
9-16	Blank Page	-
9-17	Revised	4
Back cover	Revised	-

NOTE: In case of major document revision (change of revision index letter), the document has undergone some significant changes since previous revision and all pages have been reset to the same number. All the changes and amendments introduced in the current revision, however, are marked by a vertical amendment bar so as to highlight the differences between previous and current revision of the document.

CAUTION!

This manual is valid only if it contains all of the original and revised pages listed above.

Each page to be revised must be removed, shredded and later replaced with the new, revised page in the exact same place in the manual.



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Table of contents

1 General

2 Operating limitations

3 Emergency procedures

4 Normal procedures

5 Performance

6 Weight and balance

7 Aircraft & systems

8 Handling and service

9 Appendix

**10 Supplements
(individual equipment manuals)**



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1 General



Introduction (1-2)

Technical brief (1-2)

3-view drawing (1-3)

Powerplant, fuel, oil (1-4)

Weights (1-6)

Center of gravity range (1-6)

G-load factors (1-6)

Introduction

This manual contains all information needed for appropriate and safe use of ALPHA Trainer.

IT IS MANDATORY TO CAREFULLY STUDY THIS MANUAL BEFORE USING THE AIRCRAFT.

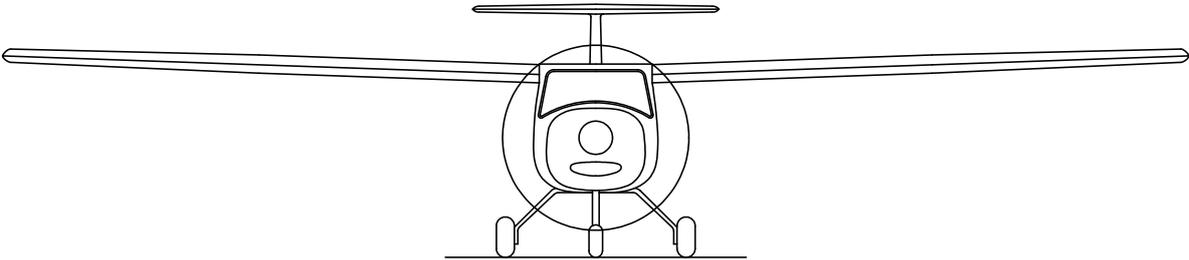
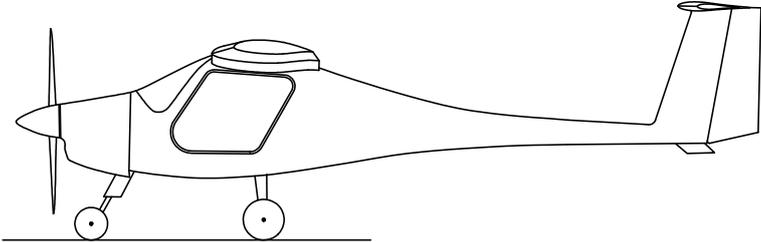
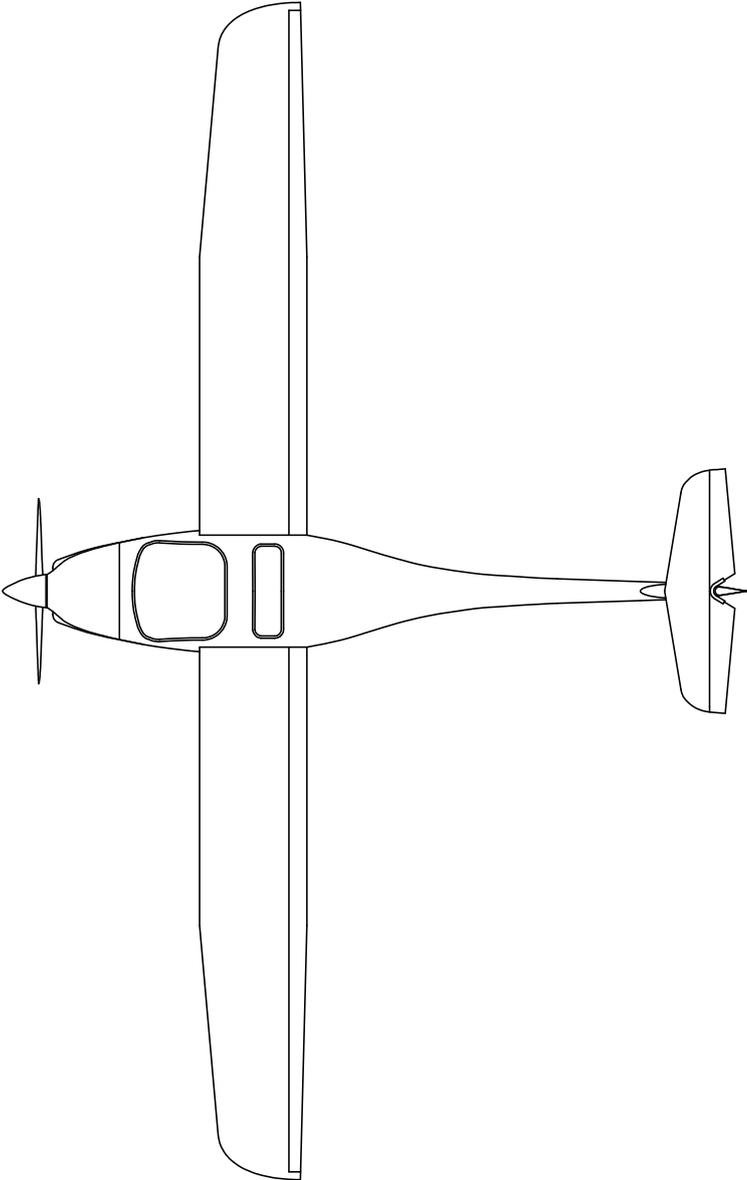
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Technical brief

DIMENSIONS	ALPHA Trainer
Wing span	34' 6" (10.5 m)
Length	21' 2" (6.45 m)
Height	6' 9" (2.05 m)
Wing surface	99.9 sqft (9.29 m ²)
Vertical fin surface	11.8 sqft (1.1 m ²)
Horizontal stabilizer and elevator surface	11.6 sqft (1.08 m ²)
Aspect ratio	11.9
Positive flap deflection (down)	15 °, 25 °
Center of gravity (MAC)	20% - 38%
Propeller - fixed pitch	Pipistrel FP02-80, dia. 65" 1660 mm

3-view drawing



Powerplant, fuel, oil

Engine manufacturer: ROTAX

Engine type: Rotax 912 (80 HP)

NOTE the parameters listed above are directly reported from OEM documentation and do not take into account possible effects of installation and engine integration with aircraft systems. Please refer to Section 2 for engine instrument markings and parameter operational ranges.

The engine

TEMPERATURE / ROTAX ENGINE	912 80 HP
Maximum coolant temp. (CT)	120 [°C]
Maximum exhaust gas temperature (EGT)	880 [°C]
Oil temperature (OIL TEMP); minimum, normal, max	50 ; 90-110 ; 140 [°C]
RPM, PRESSURE	912 80 HP
Oil pressure (OIL PRESS); lowest, highest	0.8; 2.0-5.0; 7.0 [bar]
Maximum continuous RPM	5500
maximum allowable RPM (Max 5 min)	5800
Magneto check at (RPM)	4000

Fuel and oil

ROTAX ENGINE	912 80 HP
	MOGAS min RON 90 (AKI 87)
recommended fuel**	AVGAS leaded 100LL* or unleaded Fuels with max 10% ethanol content as per Rotax specification
recommended oil**	AeroShell Oil Sport Plus 4
Oil capacity typical 3,1 quarts (3 litres)	check dipstick

*** Engine life is reduced. If using this type of fuel is unavoidable, changing the engine oil every 50 flight hours is crucial. Please consult the manufacturer on which type of oil to use.**

**** As per Rotax specification - see Service Instruction SI-912-016 "Selection of suitable operating fluids" latest issue.**

IMPORTANT!

Four-stroke engines should only be powered by unleaded fuel, for lead sedimentation inside the engine shortens its life. Provided you are unable to use unleaded fuel, make sure engine oil and the oil filter are replaced every 50 flight hours.

WARNING! Use of fuels with alcohol content exceeding 10% is not permitted.

NOTE The fuel indicator is equipped with coloured markings for fuel status which indicates the percentage of fuel tank capacity. The fuel indicator is electric and may not be reliable at all times. Pilot caution is advised.

When fuelling or de-fuelling, verify that the vent tube remains unobstructed from contamination.

Propeller

ALPHA Trainer	Propeller
Rotax 912 (80 HP)	Pipistrel FP02-80, dia. 65" 1660 mm

Weights

ALPHA Trainer weights

WEIGHT	ALPHA Trainer
Typical empty weight (refer to WBR for details)	684 lbs (310 kg)
Max. takeoff weight (MTOM)	1040 lbs (472.5 kg)
Fuel capacity (full)	13.2 USgal (50 l)
Fuel capacity (usable)	12.7 USgal (48 l)
Maximum fuel weight allowable	79.4 lbs (36 kg)
Maximum useful load (refer to WBR for details)	356 lbs (162.5 kg)
Minimum combined cockpit crew weight	121 lbs (55 kg)
Luggage weight	depends on equipment and operational configuration; maximum 22 lbs (10 kg)

WARNING! MTOM must be kept at or below 1040 lbs (472.5 kg). Pay special attention to luggage and fuel weight, as these are the applicable masses on the airframe that have an influence on center of gravity. Exceeding baggage weight limits can shift aircraft's balance to the point when the flight becomes uncontrollable! More information on baggage allowance can be found in chapter "Weight and balance".

Center of gravity range

- The aircraft's safe center of gravity position ranges between 20% and 38% of mean aerodynamic chord.
- The center of gravity point ranges between 8.7" (220 mm) and 15.0" (382 mm) aft of the datum. The datum is the wing's leading edge at the fuselage root.

G-load factors

- Max. positive wing load: + 4 G
- Max. negative wing load: - 2 G



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2 Limitations



Introduction (2-2)

Airspeed limitations (2-2)

Powerplant limitations (2-3)

Weight limits (2-4)

Cockpit crew (2-4)

Center of gravity limits (2-4)

Load factors (2-5)

Service ceiling, crosswind (2-5)

Manoeuvre limits (2-5)

Kinds of operations (2-6)

Minimum equipment list (2-6)

Other restrictions (2-7)

Placards (2-8)

Introduction

This section includes operating limitations, instrument markings and basic placards necessary for the safe operation of the airplane, its engine, standard system and standard equipment.

The limitations included in this section have been approved by target authority. Observance of these operating limitations is required by law.

Airspeed limitations

	Velocity	IAS [kts (km/h)]	Remarks
VS	Stall speed Clean	40 (74)	Stall speed flap up.
VS0	Stall speed Landing configuration	34 (64)	Stall speed flaps full.
VFE	Max. velocity flaps extended	70 (130)	Do not exceed this speed with flaps extended (+15, +25 degrees).
VA	Design manoeuvring speed	86 (160)	Do not make full or abrupt control movements above this speed.
VNE	Velocity never to be exceeded	135 (250)	Never exceed this speed in any operation.
VB	Velocity normal oper- ating	108 (201)	Maximum structural cruising speed in turbulent air.
VAE	Maximum speed with airbrakes extended	70 (130)	Do not fly with airbrakes extended above this speed.

Airspeed indicator markings

MARKING	IAS [kts (km/h)]	Definition
White band	37 - 70 (68 - 130)	Full flap operating range. Lower limit is maximum weight VS0 in landing configuration. Upper limit is maximum speed permissible with flaps extended. According to LTF-UL regulation, stall speeds on ASI are 10% higher than actual stall speeds.
Green band	43 - 108 (83 - 201)	Normal operating range lower end is maximum weight VS1 at most forward C.G. with flaps retracted. Upper limit is maximum structural cruising speed. According to LTF-UL regulation, stall speeds on ASI are 10% higher than actual stall speeds.
Yellow band	108 - 135 (201 - 250)	Maneuver the aircraft with caution in calm air only.
Red line	135 (250)	Maximum speed for all operations. VNE
Blue line	76 (140)	Best climb rate speed (Vy)
/	58 (108)	Best angle of climb speed (Vx)

Powerplant limitations

Engine manufacturer: ROTAX

Engine type: Rotax 912 (80 HP)

NOTE the parameters listed above are directly reported from OEM documentation and do not take into account possible effects of installation and engine integration with aircraft systems. Please refer to Section 2 for engine instrument markings and parameter operational ranges.

The engine

TEMPERATURE / ROTAX ENGINE	912 80 HP
Maximum coolant temp. (CT)	120 [°C]
Maximum exhaust gas temperature (EGT)	880 [°C]
Oil temperature (OIL TEMP); minimum, normal, max	50 ; 90-110 ; 140 [°C]
RPM, PRESSURE	912 80 HP
Oil pressure (OIL PRESS); lowest, highest	0.8; 2.0-5.0; 7.0 [bar]
Maximum continuous RPM	5500
maximum allowable RPM (Max 5 min)	5800
Magneto check at (RPM)	4000

Fuel and oil

ROTAX ENGINE	912 (80 HP)
recommended fuel**	MOGAS min RON 90 (AKI 87)
	AVGAS leaded 100LL* or unleaded
	Fuels with max 10% ethanol content as per Rotax specification
recommended oil**	AeroShell Oil Sport Plus 4

* **Engine life is reduced. If using this type of fuel is unavoidable, changing the engine oil every 50 flight hours is crucial. Please consult the manufacturer on which type of oil to use.**

** **As per Rotax specification - see Service Instruction SI-912-016 "Selection of suitable operating fluids" latest issue.**

IMPORTANT!

Four-stroke engines should only be powered by unleaded fuel, for lead sedimentation inside the engine shortens its life. Provided you are unable to use unleaded fuel, make sure engine oil and the oil filter are replaced every 50 flight hours.

WARNING! Use of fuels with alcohol content exceeding 10% is not permitted.

NOTE The fuel indicator is equipped with coloured markings for fuel status which indicates the percentage of fuel tank capacity. The fuel indicator is electric and may not be reliable at all times. Pilot caution is advised.

When fuelling or de-fuelling, verify that the vent tube remains unobstructed from contamination.

Propeller

ALPHA Trainer	Propeller
with Rotax 912 (80 HP)	Pipistrel FP02-80 - diameter 65" (1660 mm)

Engine instrument markings

Engine instrument markings are relevant for the pilot. These are defined by the manufacturer and may differ from the values found in OEM documentation.

Instrument	Red line (minimum)	Yellow arc (caution)	Green arc (typical)	Yellow arc (caution)	Red line (maximum)
Tachometer (RPM)	/	0-1400	1400-5500	5500-5800	5800
Exhaust Gas Temp.	650 °C (1202 °F)	750 - 750 °C (1202-1382 °F)	750 - 850 °C (1382-1562 °F)	850 - 925 °C (1562-1697 °F)	925 °C (1697 °F)
Oil temperature	50°C (122°F)	50-90°C (122-194°F)	90-110°C (194-230°F)	110-125°C (230-257°F)	125°C (257°F)
Coolant temp.	/	/	/ - 120°C (/ -248°F)	/	120°C (248°F)
Oil pressure	1 bar (15 psi)	1-1.5 bar (15-22 psi)	1.5-5 bar (22-73 psi)	5-6.5 bar (73-94 psi)	6.5 bar (94 psi)
Fuel pressure	0.15 bar (2.2 psi)	/	0.15 - 0.4 bar (2.2 - 5.8 psi)	/	0.4 bar (5.8 psi)

Weights

ALPHA Trainer weights

WEIGHT	ALPHA Trainer
Maximum takeoff weight (MTOM = MTOW)	1040 lbs (472.5 kg)
Minimum combined cockpit crew weight	121 lbs (55 kg)
Baggage area	22 lbs (10 kg) absolute limit. Always verify baggage allowance with a center of gravity calculation.

WARNING! MTOM must be kept at or below 1040 lbs (472.5 kg). Pay special attention to luggage and fuel weight, as these are the applicable masses on the airframe that have an influence on center of gravity. Exceeding baggage weight limits can shift aircraft's balance to the point when the flight becomes uncontrollable! More information on baggage allowance can be found in chapter "Weight and Balance".

Center of gravity range

- The aircraft's safe center of gravity position ranges between 20% and 38% of mean aerodynamic chord.
- The center of gravity point ranges between 8.7" (220 mm) and 15.0" (382 mm) aft of the datum. The datum is the wing's leading edge at the fuselage root.

G-load factors

Max. positive wing load: + 4 G

Max. negative wing load: - 2 G

Service ceiling, crosswind

Service ceiling is 18,000 ft. Maximum crosswind component is 18 kts.

Manoeuvre limits

ALPHA Trainer is approved as a Light Sport Aircraft and is intended for recreational and instructional flight operations. In the acquisition of various pilot certificates certain manoeuvres are required and these manoeuvres are permitted in this airplane.

The following non-aerobatic manoeuvres are permitted as defined:

- Power-on and -off stalls not below 1500 feet (450 meters) above ground level.
- Power on and off lazy eights not below 1500 feet (450 meters) above ground level, entry speed 90 kts
- Steep turns with initial speed of 80 kts.
- Chandelle manoeuvres not below 500 feet (150 meters) above ground level, entry speed 105 kts.

WARNING! Aerobatic manoeuvres, including intentional spins, are prohibited.

CAUTION! The door structure/hinge is not designed for intentional open-door operations. Be advised that the chance of door failure occurring is higher, as the airspeed at which the door is opened at increases.

Kinds of operations

ALPHA Trainer is approved for DAY - VFR operations only*. Flight into known icing conditions is prohibited.

WARNING! Should you find water drops on the airframe during pre-flight check-up at temperatures close to freezing, you may expect icing to appear in flight.

Minimum equipment list (DAY - VFR)

- Placards, checklist, this POH.
- Airspeed indicator (functional), altimeter (functional), compass (functional).
- Tachometer (RPM), CT indication (functional), OIL temp. indication (functional), OIL press. indication (functional).
- 12 V main battery (functional), alternator (functional), safety belts (2x), visual fuel indication (functional), fuel shut-off valve (functional), electric fuel (booster) pump (functional).
- Any other equipment mandated by National Authority of Registration and/or Operation.

Fuel limitations

FUEL	ALPHA Trainer
Fuel capacity (full)	13.2 USgal (50 liters)
Fuel capacity (usable - all flight conditions)	12.6 USgal (48 liters)
Unusable fuel	0.5 USgal (2 liters)
Maximum fuel weight allowable	84 lbs (38 kg)

WARNING! Takeoff is prohibited if there is less than 4 liters/1.5 USgal or when unsure about the fuel quantity on board for any reason.

WARNING! Use of fuels with alcohol content exceeding 10% is not permitted.

NOTE Maximum full capacity is indicated only through the fuel filler neck on fuselage, by visual check. At the same time, verify that the vent tube remains unobstructed from contamination.

Other restrictions

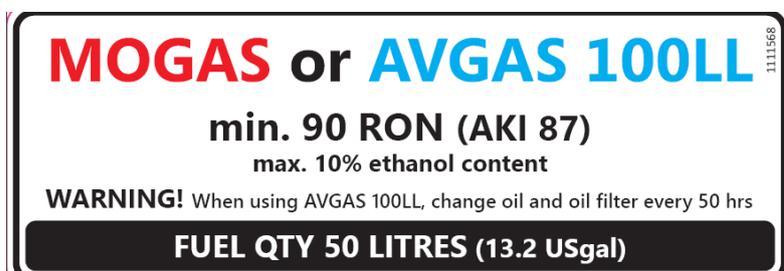
Due to flight safety reasons it is FORBIDDEN to:

- fly in heavy rainfall.
- fly during thunderstorm activity.
- fly in a blizzard.
- fly according to instrument flight rules (IFR) or attempt to fly in zero visibility conditions (IMC).
- fly when the temperature of the aircraft's surface is at risk of exceeding 55°C (130°F).
- perform aerobatic flying.
take off and land with flaps retracted.
- (landing with flaps retracted is only permitted in cases of very strong winds, but is not to be performed as a normal procedure)
- the 12 Volt power outlet is not approved to supply power to flight-critical communication or navigation devices.
- spin the aircraft.

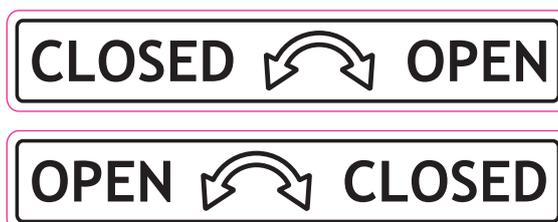
Placards

Placards (external):

Next to fuel tank filler neck:



Next to door opening latches:



Next to the fuel drain:



Next to the gascolator:
(if applicable)



Next to main wheels:



Next to nose wheel:

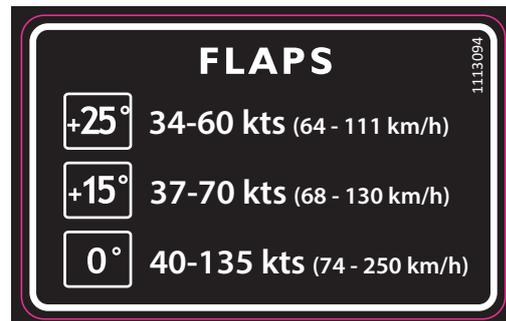
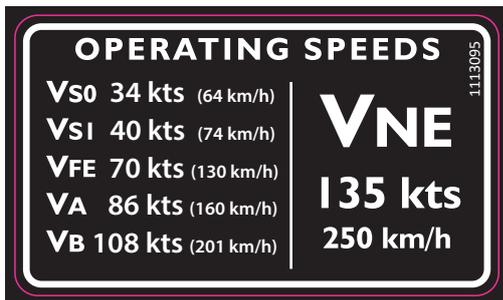


Placards (engine compartment):

On coolant bottle, oil bottle :



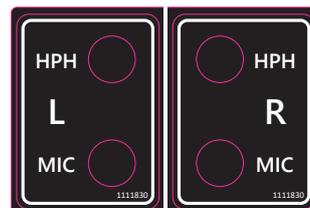
Placards (center console):



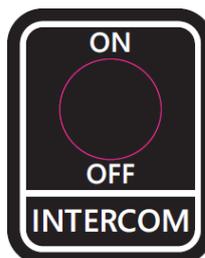
Next to microphone jacks:



or



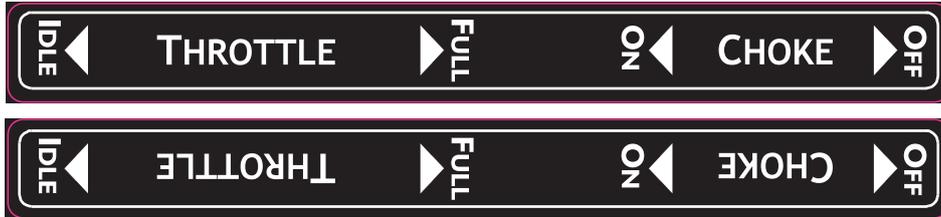
If applicable:



In front of control sticks (rudder pedal adjustment):



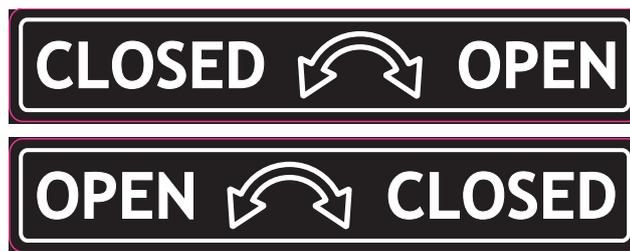
Next to choke and throttle levers:



On flap lever:



Below each door to depict door handle operation:



On upper tube in front of pilot:



Next to wheel brake pull:



Placards (Ballistic Parachute Rescue System):



**ATTENTION!
ROCKET INSIDE**

**WARNING
ROCKET FOR PARACHUTE
DEPLOYMENT INSIDE**

Next to rocket exhaust (bottom of fuselage):

**ROCKET GAS
EXHAUST**

Next to each door, top aft corner:

**This aircraft is equipped
with a rocket powered
ballistic rescue system.**

Next to activation handle (cockpit)

**PULL FOR PARACHUTE
DEPLOYMENT**

On the instrument panel:





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3 Emergency procedures



Introduction (3-2)

Stall recovery (3-2)

Spin recovery (3-2)

Engine failure (3-3)

**Emergency landing /
Outlanding (3-3)**

Engine fire (3-3)

Smoke in cockpit (3-4)

Carburettor icing (3-4)

Electrical system failure (3-5)

Flutter (3-5)

Exceeding VNE (3-5)

Ditching (3-5)

Icing / Pneumatic failure (3-5)

Airbrake failure (3-5)

Introduction

This section provides information for coping with emergencies that may occur. Emergencies caused by airplane or engine malfunctions are extremely rare if proper pre-flight inspections and maintenance are practiced. En route weather emergencies can be minimized or eliminated by careful flight planning and good judgment when unexpected weather is encountered. In any emergency, the most important task is continued control of the airplane and manoeuvring to execute a successful landing.

Stall recovery

First reduce the angle of attack by pushing the control stick forward, then

- 1. Reduce angle of attack by releasing stick to neutral position.**
- 2. Add full power (throttle lever in full forward position).**
- 3. Resume horizontal flight.**

Spin recovery

ALPHA Trainer is constructed in such a manner that it is difficult to be flown into a spin, and even so only at aft center of gravity positions. However, once spinning, react as follows:

- 1. Set throttle to idle (lever in full back position).**
- 2. Apply full rudder deflection in the direction opposite the spin.**
- 3. Lower the nose towards the ground to build speed (release stick).**
- 4. As the aircraft stops spinning neutralise rudder deflection.**
- 5. Slowly pull up and regain horizontal flight.**

WARNING! Keep the control stick centred along its lateral axis (no aileron deflections throughout the recovery phase! Do not attempt to stop the aircraft from spinning using ailerons instead of rudder!

WARNING! After having stopped spinning, recovering from the dive must be performed using gentle stick movements (pull), rather than overstressing the aircraft. However, VNE must not be exceeded during this manoeuvre.

Resume normal flight when the aircraft is straight and level.

Engine failure

Engine failure during takeoff

Ensure proper airspeed first (100 km/h / 55 kts) and land the aircraft on the runway heading, avoiding obstacles in your way. Set fuel shut-off valve to OFF and set master / battery switch to OFF position .

WARNING! DO NOT CHANGE COURSE OR MAKE TURNS IF THIS IS NOT OF VITAL NECESSITY!
After having landed safely, ensure protection of aircraft and vacate the runway as soon as possible to keep the runway clear for arriving and departing traffic.

Rough engine operation or engine failure in flight

First ensure proper airspeed (120 km/h / 64 kts), then start analysing terrain underneath and choose the most appropriate runway or site for landing out.

Provided the engine failed aloft, react as follows:

Make sure the master / battery switch is in the ON position, magneto switches both set to ON, fuel valve OPEN and Fuel (Booster) pump ON. Attempt to restart the engine. If unsuccessful, begin with the landing out procedure immediately.

Emergency landing / Outlanding

1. Shut off fuel valve.
2. Fuel (Booster) pump OFF.
3. Magneto L/R - OFF.
4. Seatbelts - secured.
5. Approach and land with extreme caution, maintaining normal airspeeds.
6. Radio: transmit Mayday - Transponder: squawk 7700 (if time permits).
7. ELT (if installed) switch ON (if necessary, especially when over remote/terrainy areas)
8. Flaps (+2) - when landing is assured.
9. Master / battery OFF - just before touch down (flight instrument will switch off).
10. After having landed leave the aircraft immediately.

The outlanding manoeuvre MUST be preformed with regard to all normal flight airspeed parameters.

Engine fire

Engine fire on ground

This phenomenon is very rare in the field of sport aviation. However, if an engine fire occurs on ground, react as follows:

1. Shut fuel valve (OFF).
2. Fuel (Booster) pump OFF.
3. Come to a full-stop, engage starter and set throttle to full power (lever full forward).
4. Disconnect the battery from the circuit (pull battery disc. ring on the switch column).
5. Master / battery switch OFF immediately after the engine has stopped.
6. Abandon the aircraft and start the fire extinguishing.

WARNING! After the fire has been extinguished DO NOT attempt to restart the engine.

Engine fire in flight

1. Shut fuel valve OFF.
2. Fuel (Booster) pump OFF.
3. Set full power (throttle lever in full forward position) until engine stops.
4. Set magnetos to OFF.
5. Keep avionics / generator ON and master / battery ON as required, on approach set both OFF.
6. Set ventilation for adequate breathing. Keep in mind that oxygen intensifies fire.
7. Perform side-slip (crab) manoeuvre in direction opposite the fire.
8. Perform emergency landing out procedure.

Smoke in cockpit

Smoke in cockpit is usually a consequence of electrical wiring malfunction. As it is most definitely caused by a short circuit it is required that the pilot reacts as follows:

1. Fuel (Booster) pump switch OFF.
2. Avionics / generator OFF.

WARNING! Setting Avionics / generator switch off will switch off all instruments but enables unobstructed engine operation while at the same time disconnects all other electrical devices from the circuit. Verify that the 12 V socket is OFF as well.

3. Disconnect the battery from the circuit (pull battery disconnection ring on the instrument panel's switch column).
4. Land as soon as possible.

In case you have trouble breathing or the visibility out of the cockpit has degraded severely due to the smoke, open the cabin door and leave it hanging freely. Flying with the door open, do not, under any circumstances exceed 110 km/h / 60 kts.

Carburettor icing

First noticeable signs of carburettor icing are rough engine noises and gradual loss of power.

Carburettor icing may occur even at temperatures as high as 10°C (50°F), provided the air humidity is increased.

Should you be suspecting carburettor icing to take place, descend immediately into warmer and/or less humid air!

In case of complete power loss perform emergency landing out procedure.

Electrical system failure

The engine will continue to function due to the onboard alternator and battery. In case of battery failure, be aware that the engine can keep running, however a re-start will not be possible. In event of alternator failure, the battery will support the onboard avionics. In event of double power source failure, use analogue on-board instruments and land normally.

Flutter

Flutter is defined as the oscillation of control surfaces. It is in most cases caused by abrupt control deflections at speeds close to or in excess of VNE. As it occurs, the ailerons, elevator or even the whole aircraft start to vibrate violently.

Should flutter occur, increase angle of attack (pull stick back) and reduce throttle immediately in order to reduce speed and increase load (damping) on the structure.

WARNING! Fluttering of ailerons or tail surfaces may cause permanent structural damage and/or inability to control the aircraft. After having landed safely, the aircraft **MUST** undergo a series of check-ups performed by authorized service personnel to verify airworthiness. Inform the manufacturer about the incident.

Exceeding VNE

Should the VNE be exceeded, reduce airspeed slowly and continue flying using gentle control deflections. Land safely as soon as possible and have the aircraft verified for airworthiness by authorised service personnel.

Ditching

Should you be forced to land in a body of water, use the same emergency procedure as above for the "Emergency landing / Landing out" case. In addition, make sure to open both doors fully before hitting the water, as well as disconnect the battery from the circuit (pull ring on electrical panel). Touch the water with the slowest possible speed, possibly from a high-flare situation.

Icing/Pneumatic instrument failures

Turn back or change altitude to exit icing conditions. Consider lateral or vertical path reversal to return to last "known good" flight conditions. Maintain VFR flight!

Set cabin heating ON. Watch for signs of icing on the pitot tube. In case of pneumatic instrument failures, use the GPS (optional) information to reference to approximate ground speed. Plan the landing at the nearest airport, or a suitable off airport landing site in case of an extremely rapid ice build-up. Increase the speed to avoid stall.

Manoeuvre the airplane gently and leave the wing flaps retracted. When ice is built up at the horizontal stabilizer, the change of pitching moment due to flaps extension may result of loss of elevator control. Approach at elevated speeds (70 kts, also if using the GPS as a reference).

WARNING! Failure to act quickly may result in an unrecoverable icing encounter.

Airbrake failure

Should the airbrakes fail to extend/retract due to mechanical or electrical issues, the STATUS LED on the instrument panel will begin blinking (2 blinks per second). In this case:

Maintain speeds of 70 KIAS / 130 km/h or below and land when possible. You can attempt to regain control of the airbrakes by manually switching between the two settings EXT and RETR. Be aware that the approach path needs to be adjusted if the airbrakes remain retracted/extended.

NOTE If a single airbrake failure occurs, the system will position the working airbrake to match the degree of extension, eliminating the asymmetric condition.



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4 Normal procedures



Daily inspection (4-2)

Pre-flight inspection (4-2)

**Normal procedures and
recommended speeds (4-5)**

Daily inspection

The daily inspection matches the pre-flight inspection.

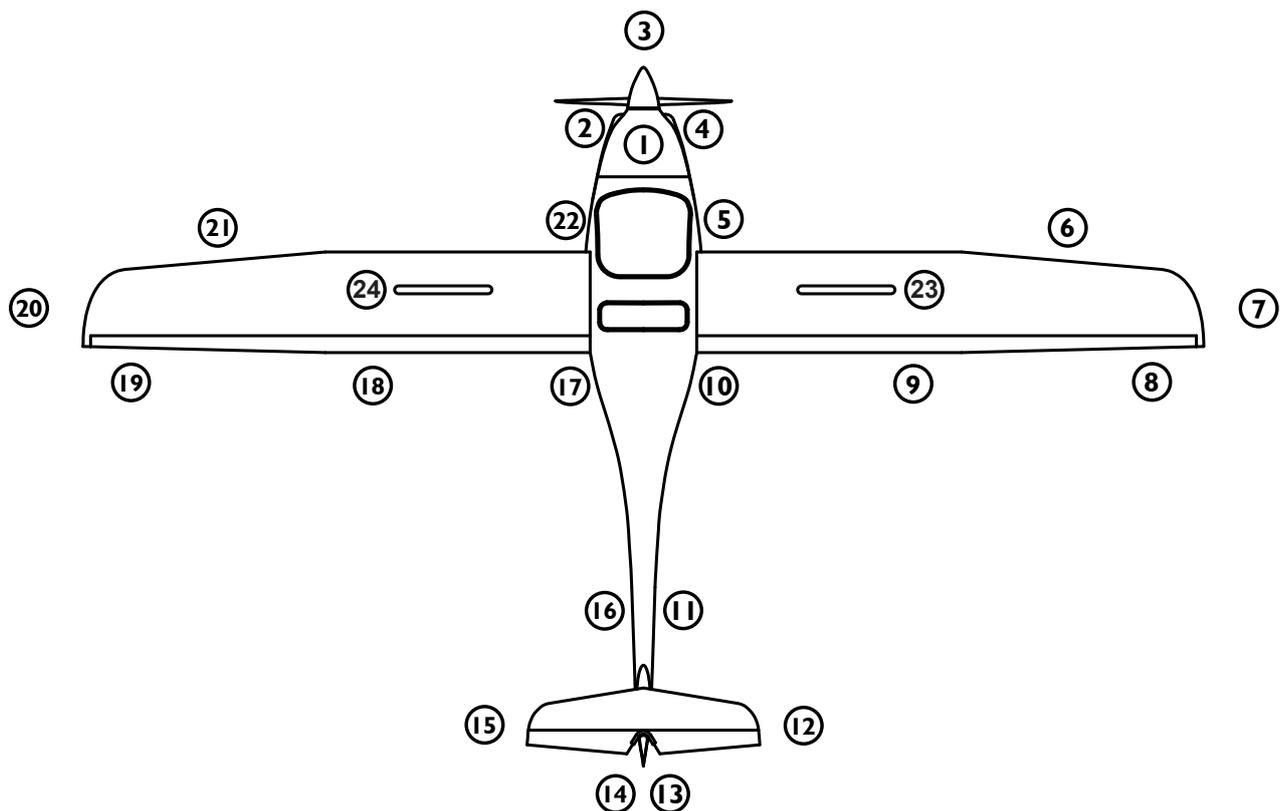
Pre-flight inspection

WARNING! Every single inspection mentioned in this chapter must be performed prior to EVERY FLIGHT, regardless of when the previous flight took place!

The person responsible for the pre-flight inspection is the pilot, who is required to perform the check-up in the utmost thorough and precise manner.

If the status of any of the parts and/or operations does not comply with conditions stated in this chapter, the damage **MUST** be repaired prior to engine start-up. Disobeying these instructions may result in serious additional damage to the plane and crew, including injury and loss of life!

Schematic of pre-flight inspection



- | | | |
|------------------------------|--------------------------------|---------------------------------|
| 1 Engine, engine cover | 9 Right wing - continued | 17 Fuselage (LH side) |
| 2 Gascolator | 10 Fuselage (RH side) | 18 Left wing - continued |
| 3 Spinner, Nose wheel | 11 Fuselage, continued (right) | 19 Left wing - trailing edge |
| 4 Propeller | 12 Hor. tail surfaces (right) | 20 Left wingtip, lights |
| 5 Undercarriage, RH wheel | 13 Vert. tail surfaces (right) | 21 Left wing - leading edge |
| 6 Right wing - leading edge | 14 Vert. tail surfaces (left) | 22 Undercarriage, LH wheel |
| 7 Right wingtip, lights | 15 Hor. tail surfaces (left) | 23 Electric airbrakes (LH side) |
| 8 Right wing - trailing edge | 16 Fuselage, continued (left) | 24 Electric airbrakes (RH side) |

Engine, engine cover ①

Cooling fluid level: half way to the top

Oil quantity: within designated limits

Throttle, choke and oil pump wires: no mechanical damage, smooth and unobstructed movement

Radiators and hoses: no mechanical damage and/or leakage, air filters clean and intact

Exhaust pipes and muffler: firmly in position, no cracks, springs intact and in position, rubber dumpers intact

Fuel and/or oil leakage: no fluid on hoses, engine housing or engine cover

Reduction gearbox: check for eventual oil leakage, all bolts and plugs attached firmly

Fasteners and engine cover screws: tightened, engine cover undamaged

Gascolator ②

Drain approximately 1 cup of fuel and check for contamination.

Spinner ③

Spinner: no mechanical damage (e.g. cracks, impact spots), screws tight

Bolts and nuts: secured

Nose wheel: grab aircraft's propeller and push it towards the ground to verify proper nose wheel suspension operation. Then lift the nose wheel off the ground and check for nose leg strut free play.

Bolts: fastened

Tire: no cracks, adequate pressure

Wheel fairing: undamaged, firmly attached, clean (e.g. no mud or grass on the inside)

Propeller ④

Hub and blades: no mechanical damage (e.g. cracks), both immaculately clean

Bolts and nuts: secured

Undercarriage, wheels ⑤ ②②

Bolts: fastened

Landing gear strut: no mechanical damage (e.g. cracks), clean

Wheel: no mechanical damage (e.g. cracks), clean

Wheel axle and nut: fastened

Fluid line (hydraulic brakes): no mechanical damage and/or leakage

Tire: no cracks, adequate pressure

Wheel fairing: undamaged, firmly attached, clean (e.g. no mud or grass on the inside)

Wing leading edge ⑥ ②①

Surface condition: pristine, no cracks, impact spots, no paint and/or edge separations

Pitot tube: firmly attached, no mechanical damage or bends. Remove protection cover and make sure it is not blocked or full of water.

Wing drain holes: make sure they are not blocked and clean accordingly.

Wingtip, lights ⑦ ⑳

Surface condition: pristine, no cracks, impact spots or bumps, no paint separations

Wings' trailing edge ⑧ ⑲

Surface condition: pristine, no cracks, impact spots, no paint and/or edge separations

Aileron: pristine surface, no cracks and/or impact spots, no paint abnormalities and edge separations, no vertical or horizontal free play, smooth and unobstructed deflections

Fuel reservoir cap ⑨ ⑱

Fuel reservoir cap: fastened. Make sure the vent pipe is completely clean.

Fuselage, antenna, rescue parachute cover ⑩ ⑰

Self-adhesive tape: in position, no separations

Flaperon control system cover, antenna: firmly attached

Fuselage, continued ⑪ ⑯

Surface condition: pristine, no cracks, impact spots or bumps, no paint separations

Horizontal tail surfaces ⑫ ⑮

Surface condition: pristine, no cracks, impact spots or bumps, no paint and/or edge separations

Hinges: no free play in any direction

Horizontal stabilizer attachment mechanism: fastened and secured

Mylar seals covering the gap between horizontal and vertical tail surfaces: in position

Elevator: smooth and unobstructed up-down movement, no side-to-side free play

Vertical tail surfaces ⑬ ⑭

Vertical fin bottom part: no cracks, impact spots or paint separations along main chord

Surface condition: pristine, no cracks, impact spots or bumps, no paint separations

Hinges: no free play in any direction

Rudder cable endings: intact, bolts in position

Electric airbrakes ⑳㉔

Verify that the airbrakes are in the closed position and flush with the upper wing surface.

CAUTION! Pre-flight inspection should be performed following stations 1 through 24!

Cockpit pre-flight inspection

Instrument panel and instruments: checked

Battery disconnection lever: in position for battery operation (lever deflected towards the firewall)

Master / battery switch OFF: no control lights and/or electronic instrument activity

Fuel (Booster) pump: OFF

Master / battery switch ON: generator fail light on (engine not running!)

Make sure you have set all instruments to correct initial setting.

Main wing spars and connectors: no visible abnormalities of metal parts, spars, pins and bolts; all bolts and nuts in position and tightened

Fuel hoses, pitot-static lines and electrical cables: correctly connected and in position

Safety belts: undamaged, verify unobstructed harness opening; fastening points intact

All accessible electrical harnesses, fuel system components, pitot-static lines and control system components behind the cabin*: Remove Seats, CHECK condition, VERIFY no fuel odour.

All electrical, fuel and pitot-static wing connections: CHECK condition

All visible rudder cables: CHECK condition

Doors and windshield: perfect closing at all three points, smooth opening, hinges firmly attached; transparent surfaces immaculately clean with no cracks.

Electric airbrakes: Perform calibration procedure (see chapter 7). Select AUTO with the airbrakes control knob and extend the flaps fully (second stage). Verify that the airbrakes status LED is illuminated (solid, no blinking). Retract the flaps fully and check that the status LED turns off.

Flap handle: button spring firm, locking mechanism working properly, smooth movement along full deflections, no free play or visible damage.

Radio wiring: test the switches, check connectors and headset, perform radio check

Battery (some models): firmly in position, joints clean with wires connected

Emergency parachute release handle (optional): safety pin removed.

Elevator trim: verify travel and set neutral before start-up.

Make sure unobstructed access is provided.

ELT (transmitter and remote switch) (if installed): Arm/Armed

If installed, check that ELT transmitter and remote switch are BOTH in the ARM/OFF (armed) position. Antenna and cable harness shall be properly connected to ELT transmitter! Periodical testing of ELT operation is required (advised once a month, but not more than once per week). Please see OEM for details about testing procedure.

* Only before first flight of the day.

CAUTION! Make sure that baggage compartment limitations (22 lbs - 10 kg max) and aircraft weight and balance limitations are respected.

Normal procedures, recommended speeds

To enter the cabin first lift the door all the way to the bottom wing surface. The silver knob will grab and secure the door in position. Sit on the cabin's edge and grab it to support your body. Drag yourself onto the seat, lifting only one leg over the stick for best position. Immediately after positioning yourself in the seat, check the rudder pedals' position suits your size and needs. The pedals' position is also adjustable during flight.

To close the door, gently pull the silver knob and allow the door to fall into the vertical position. Check for obstructions at the lower and side connection points. With a gentle inwards pressure, rotate the handle into the over center position (slightly past horizontal).

CAUTION! Failure to fully rotate handle can cause inadvertent door opening in flight due to aircraft vibration and result in door damage.

CAUTION! If a door needs to be opened in flight, (for example to retrieve a seatbelt strap), firmly grasp the handle so as to not allow the door to raise uncontrolled up into the airflow.

Fasten the safety belts according to your size.

Adjust the rudder pedals according to your required legroom. Sit inside the cockpit and release the pressure off the pedals. Pull the black knob in front of the control stick to bring the pedals closer to

you. To move the pedals further away, first release the pressure of the pedals, then pull on the knob slightly (this will release the lock in the mechanism). Now push the pedals forward using with your feet, while keeping the black adjustment knob in your hand.

WARNING! The safety harness must secure you in your seat. This is especially important when flying in rough air, or else you may bump your head on the cabin support strut assembly and/or wing spars. Make sure you tighten the bottom straps first, then shoulder straps.

Engine start-up

Before engine start-up

CAUTION! To ensure proper and safe use of the aircraft it is essential to familiarize yourself with engine's limitations and engine manufacturer's safety warnings. Before engine start-up make sure the area in front of the aircraft is clear. It is recommended to start-up the engine with the aircraft's nose pointing into the wind.

Make sure the fuel quantity is sufficient for the planned duration of flight.
Make sure the pitot tube is uncovered and rescue parachute safety pin removed.
Engage wheel brakes. If equipped with the parking brake, engage parking brake.

Engine start-up

Make sure fuel valve is OPEN and master/battery switch in OFF position, avionics / generator switch OFF.

Should the engine be cold, apply choke (lever full back).

Set master / battery switch ON, then set avionics / generator switch ON, set both magnetos ON, engage engine starter and keep it engaged until the engine starts. Verify fuel pressure within limits. Set throttle to 2500 RPM.

Slide the choke lever forward gradually.

CAUTION! When the engine is very cold, the engine may refuse to start. Should this occur, move the choke handle fully backwards and hold it there for some 20 seconds to make mixture richer.

Engine warm-up procedure

The engine should be warmed-up at 2500 RPM up to the point working temperature of oil is reached.

Warming-up the engine you should:

- 1 Point aircraft's nose into the wind.
- 2 Verify the engine temperature ranges within operational limits.

CAUTION! Avoid engine warm-up at idle throttle as this causes the spark plugs to turn dirty and the engine to overheat.

After the working oil temperature has reached, engage wheel brakes and control stick in full back position, first set engine power to 4000 RPM in order to perform the ignition check. Set the ignition switches OFF and back ON one by one to verify RPM drop of not more than 500 RPM and the difference in drop should not exceed 150 RPM.

When the ignition check has been completed, add full power (throttle lever full forward) and monitor engine's RPM. Make sure they range between maximum recommended and maximum allowable RPM limits.

WARNING! The RPM drop must not exceed 500 RPM for either “magneto” and the difference in drop should not exceed 150 RPM.

NOTE The RPM drop must not exceed 500 RPM for either “magneto” and the difference in drop should not exceed 150 RPM. If there is doubt concerning operation of the ignition system, RPM check at higher engine speeds will usually confirm whether a deficiency exists. An absence of RPM drop may indicate faulty grounding of one side of the ignition system or magneto timing set in advance of the specified setting.

NOTE The engine does not reach 5800 RPM on ground. Engines are factory set to reach maximum ground RPM of 5300 - 5500 at sea level at 25 degrees Celsius. Maximum ground RPM may vary depending on the season and service elevation.

CAUTION! Should engine’s RPM be lower than the recommended on ground amount (min. 5000 RPM) or in excess of maximum allowable RPM on ground (5800) during this manoeuvre, check engine and wiring for correct installation.

Taxi

Release parking brake if set and release the handbrake. Taxiing technique does not differ from other aircraft equipped with a steerable nose wheel. Prior to taxiing it is essential to check the wheel brakes for proper braking action.

In the case you expect taxi a long way, take engine warm-up time into account and begin taxiing immediately after engine start-up. Warm-up the engine during taxiing so the engine does not overheat because of prolonged ground operation.

Holding point

Make sure the temperatures at full power range are within operational limits.

Make sure the safety belts are fastened and doors closed and secured at all three closing points.

For short field operations set flaps to the 2nd position (+25 degrees, flap handle full up). For all other operations select 1st position (+15°). Ensure the airbrakes are closed for takeoff.

Power idle.

Fuel (Booster) pump: ON

Fuel pressure: check (increase)

CAUTION! Should the engine start to overheat because of long taxi and holding, shut down the engine and wait for the engine temperatures drop to reasonable values. If possible, point the aircraft’s nose into the wind. This will provide radiators with airflow to cool down the engine faster.

Electric airbrake use

Typically the airbrakes are used in AUTO mode. In this mode, they extend when the flaps are extended to 25°, which causes a 250-300 fpm increase in sink rate. They retract as soon as another flap position is chosen.

CAUTION! In AUTO mode, modify the take-off procedures to use flaps extended to 15°. In case of short field takeoffs (flaps extended to 25°), AUTO mode must not be engaged and the control knob must be used to manually retract the airbrakes.

Take-off and initial climb

Before lining-up verify the following:

Parking brake / brakes : disengaged (full forward)

Fuel valve: fully open

Fuel quantity: sufficient

Safety belts: fastened

Cabin doors: closed securely

Trim handle: in neutral position or slightly forward

Flap handle: for short field operations set flaps to the 2nd position (+25 degrees, flap handle full up) and airbrakes RETR. For all other operations select 1st position (+15°).

Ensure the airbrakes are fully retracted.

Runway: clear

Now release brakes, line up and apply full power.

Verify engine RPM at full throttle (5300 - 5500 RPM).

CAUTION! Keep adding power gradually.

WARNING! Should engine RPM not reach more than 5000 RPM when at full throttle, **ABORT TAKE-OFF IMMEDIATELY, come to a standstill and verify systems.**

Start the takeoff roll pulling the control stick one third backward and lift the nose wheel off the ground as you accelerate. Reaching 40-43 kts (75-80 km/h), gently pull on the stick to get the aircraft airborne.

CAUTION! Crosswind (max 18 kts) takeoff should be performed with control stick pointed into the wind. Special attention should be paid to maintaining runway heading!

Initial climb

When airborne, engage brakes momentarily to prevent in-flight wheel spinning.

Accelerate at full power and later maintain proper climbing speed.

As you reach 60 kts (110 km/h) above 150 ft (50 m), set flaps to 1st stage, reaching 70 kts (130 km/h) at 300 ft (100 m) retract flaps. Reduce RPM by 10% or below 5500 RPM and continue climbing at 76 kts (140 km/h).

Adjust the trim to neutralize the stick force if necessary.

Remember to keep the temperatures and RPM within operational limits during climb out.

CAUTION! Reduce power and lower nose to increase speed in order to cool the engine down if necessary.

Should you be climbing for a cross-country flight, consider climbing at 100 kts (185 km/h) as this will greatly increase your overall travelling speed.

Reaching cruise altitude, establish horizontal flight and set engine power to cruise (5300 RPM).

Cruise

As horizontal flight has been established, verify on-board fuel quantity again.

Fuel (Booster) pump OFF.

Keep the aircraft balanced while maintaining desired flight parameters.

Check engine operation and flight parameters regularly! Recommended cruise is at 5300 RPM, with an expected fuel burn of around 13.6 lph (3.6 USgal) per hour.

Cruising in rough conditions

Should you experience turbulence, reduce airspeed below VB and continue flying with flaps retracted.

CAUTION! In rough air, reduce engine power if necessary to keep airspeed below VB.

Descent and final approach

Descent at speeds at or below VB and flaps retracted.

For approach reduce speed to 70 kts (130 km/h) and set flaps to 1st position before turning to base leg. Set Fuel (Booster) pump to ON.

Adjust engine power to maintain proper airspeed. Set trim to neutralise stick force if necessary.

During the descent monitor temperatures and keep them within operational limits.

CAUTION! During the descent engine power **MUST** be reduced. Should you be forced to descend at idle power, make sure you keep adding throttle for short periods of time, not to turn the spark plugs dirty.

On final, set flaps to 2nd position.

Select electric airbrakes if required.

Align with the runway and reduce power to idle.

Maintain an airspeed of 55 kts (102 km/h).

Use throttle to control your descent glide path, otherwise control your attitude and crab if necessary.

NOTE: When landing with retracted flaps (high winds, strong turbulence), increase airspeed for half-wind-gust intensity.

CAUTION! Crosswind landings require higher final approach speeds to ensure aircraft's safe manoeuvrability. Increase the approach speed by 1 kts for every 1 kts of crosswind component e.g. in case of 5 kts crosswind component, increase the approach speed by 5 kts.

Roundout and touchdown

CAUTION! See chapter "Performance" for landing performance.

Roundout and touchdown (flare) occurs at following airspeeds:

Calm air, aircraft at MTOM	40 kts (75 km/h) IAS
Rough air, aircraft at MTOM (incl. strong crosswinds up to 34 km/h (18 kts))	42 kts (78 km/h) IAS

CAUTION! Land the aircraft in such a manner that the two main wheels touch the ground first, allow the nose-wheel touchdown only after speed has been reduced below 25 kts. When lowering the nose wheel to the runway the rudder **MUST NOT** be deflected in any direction (rudder pedals centred).

When on ground, start braking action holding the control stick in full back position. Steer the aircraft using rudder only. Provided the runway length is sufficient, come to a complete standstill without engaging the brakes holding the control stick slightly backwards as you slow down.

NOTE: In event of hard landing ELT may activate if installed. Transmission is signalled by the blinking LED light on the ELT remote switch. Stop the transmission by resetting the switch (see applicable OEM for details).

Crosswind approach and roundout

CAUTION! Crosswinds prolong landing runway length due to elevated airspeed that should be used, see previous page.

Performing a crosswind landing, the wing-low method should be used. When using the wing-low method it is necessary to gradually increase the deflection of the rudder and aileron to maintain the proper amount of drift correction.

WARNING! If the crab method of drift correction has been used throughout the final approach and roundout, the crab must be recovered before the touchdown by applying rudder to align the aircraft's longitudinal axis with its direction of movement.

Parking

Come to a complete standstill by engaging the wheel brake.
Re-check RPM drop by switching ignition OFF and back ON, one by one.
Set Fuel (Booster) pump OFF.
Leave the engine running at idle RPM for a minute in order to cool it down.
Retract flaps and airbrakes.
Set avionics / generator switch OFF, all green switches OFF, then both magnetos OFF. Master / battery switch OFF.

Insert parachute rescue system handle's safety pin. Apply parking brake. Open cabin door, unfasten safety belts and exit the cockpit. Chock the wheels and secure the pitot tube by putting on a protection cover.

It is recommended to shut the fuel tank valve.

Restarting the engine in flight

This procedure applies only for restarting the engine following an intentional unpowered flight.

Reduce speed to 50 kts (90 km/h).
Apply normal engine start-up procedure.

Engine restart in flight is possible up to service ceiling altitude of 18,000 ft.

Should the engine cool down during unpowered flight, apply choke. Always start the engine at idle throttle.

CAUTION! Do not add full power while the engine is still cool. Fly at lower airspeeds at low power engine setting to warm it up instead (e.g. 50 kts (90 km/h) at 3000 RPM).

5 Performance



Introduction (5-2)

Airspeed indicator calibration (5-2)

Take-off performance (5-2)

Climb performance (5-4)

Cruise (5-5)

Descent (5-5)

Landing performance (5-6)

Crosswind takeoffs/landings (5-6)

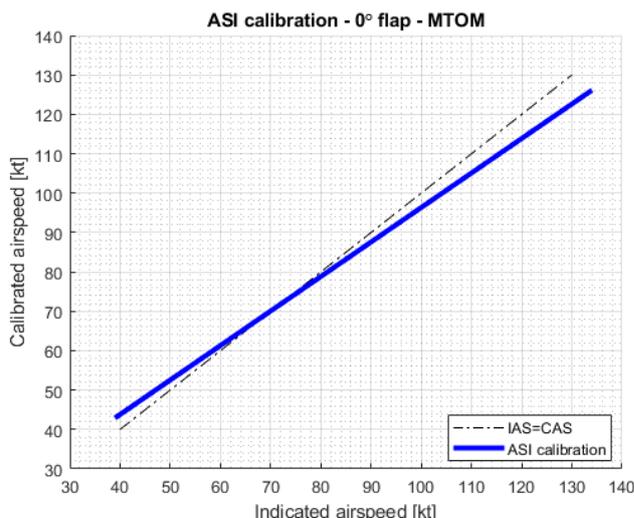
Introduction

This section provides information about the aircraft's airspeed calibration, stall speeds and general performance. All data published was obtained from test flight analysis using average flying skills.

ALPHA Trainer has demonstrated adequate engine cooling performance at ambient temperatures of 45°C (113 F). This is not to be regarded as the limit temperature, however temperatures higher than the mentioned may have an adverse effect on engine cooling and overall performance.

Airspeed indicator calibration (IAS to CAS)

The IAS-CAS relation is shown in the graph below:



Stall speeds

Stall speeds at MTOM (1040 lbs / 472.5 kg) for ALPHA Trainer are as follows:

Flaps retracted; 0° (retracted):	40 kts (74 km/h)
Flaps in 1 st position; +15° (down):	37 kts (68 km/h)
Flaps in 2 nd position; +25° (down): (without airbrakes)	34 kts (64 km/h)

Take-off performance

All data, unless otherwise noted, published in this section was obtained under following conditions:
 aircraft at MTOM
 elevation: sea level
 wind: calm
 runway: grass runway
 Data extrapolated for ICAO standard atmosphere

ALPHA Trainer	ALPHA Trainer
Takeoff ground roll at MTOM	555 ft (169 m)
Takeoff runway length (over 50 ft/15 m obstacle)	870 ft (265 m)

NOTE In order to meet the data for takeoff runway length over 50 ft obstacle maintain V_x (52 kts, 98 km/h) after take-off.

Takeoff runway length may vary depending on the wind, temperature, elevation and wing & propeller surface condition.

Effect of elevation

The table below provides data about the effect of elevation on takeoff runway length.

Elevation (m)	0	500	1000	1500
Atmosph. pressure (inHg)	29.92	28.17	26.52	24.95
Atmosph. pressure (hPa)	1013	954	898	845
Outside temperature (°C)	15,0	11,7	8,5	5,2
Outside temperature (°F)	59	53	47	41
Takeoff ground roll				
ALPHA Trainer	555 ft (169 m)	700 ft (213 m)	870 ft (265 m)	1090 ft (332 m)
Takeoff distance over 50 ft / 15 m obstacle				
ALPHA Trainer	870 ft (265 m)	1035 ft (315 m)	1295 ft (395 m)	1420 ft (433 m)

WARNING: Calculating take-off performance in hot atmosphere, it is mandatory to consider the takeoff runway length prolongs as follows: $L = 1,10 \cdot (L_h + L_t - L_0)$.

Abbreviations are as follows:

L_h = takeoff runway length at present elevation, ISA standard conditions

L_t = takeoff runway length at sea level at same temperature/wind as on the given location,

L_0 = zero wind takeoff runway length at 15°C at sea level.

e.g. if outside temperature is 25°C and you are on 500 m elevation, your takeoff runway length will be: $L = 1,10 \cdot (L_h + L_t - L_0) = 1,10 \cdot (205 \text{ m} + 215 \text{ m} - 180 \text{ m}) = 264 \text{ meters}$.

Effect of the wind

Wind (head, cross or tailwind) affects aircraft's ground speed (GS).

Headwind on takeoff or landing causes the takeoff and landing distance length to shorten as the GS is smaller during these two flight stages. The opposite holds true for tailwind on takeoff and landing as tailwind prolongs takeoff and landing distances significantly.

Headwind shortens takeoff and landing distances by 25 feet (8 meters) with every 3 kts (5 km/h) of wind increase (e.g. provided there is a 6 kts (10 km/h) headwind on takeoff and landing, distances will be approximately 50 ft (16 meters) shorter than ones published in the manual).

Tailwind prolongs takeoff and landing distances by 60-65 feet (18-20 meters) with every 3 kts (5 km/h) wind increase (e.g. provided there is a 6 kts (10 km/h) tailwind on takeoff or landing, distances will be approximately 120-130 feet (36-40 meters) longer than ones published in the manual).

WARNING! Tailwind affects takeoff and landing performance by more than twice as much as headwind does.

The table below provides data about the effect of headwind (+) and tailwind (-) on takeoff runway length (referenced for sea level conditions, airplane at MTOM). Relative effect is maintained at any elevation.

Windspeed (kts)	-6	-4	-2	0	4	8	12
Takeoff runway length [ft]							
ALPHA Trainer	680	645	605	555	525	495	480
Takeoff distance over 50 ft / 15 m obstacle [ft]							
ALPHA Trainer	1130	1065	965	870	810	760	720

Effect of outside air temperature

The table below provides data about the effect of outside temperature on takeoff runway length. Data is referenced for sea level performance at MTOM.

Temperature (°C)	15	20	25	30	35
Takeoff runway length [ft]					
ALPHA Trainer	555	670	820	935	1020
Takeoff distance over 50 ft / 15 m obstacle [ft]					
ALPHA Trainer	870	965	1045	1280	1410

extreme outside temperature performance

Temperature (°C)	0	10	13	40	45
Takeoff runway length [ft]					
ALPHA Trainer	455	490	535	1110	1040
Takeoff distance over 50 ft / 15 m obstacle [ft]					
ALPHA Trainer	785	835	850	1490	1640

Climb performance

ALPHA Trainer	ALPHA Trainer
Best climb speed	54 kts (100 km/h)
Best climb rate at MTOM, sea level	1220 fpm (6.1 m/s)
Climb rate at 100 kts (185 km/h), sea level	800 fpm (4.0 m/s)

Effect of outside air temperature

For every 5 degrees Celsius (10 F) of OAT increase versus the ISA, the climb rate decreases by 60 fpm (0.3 m/s).

Effect of altitude

The table below provides data about the effect of elevation on climb rate at best climb speed V_y at MTOM

ALPHA Trainer	ALPHA Trainer
0 m (0 ft)	1220 fpm (6.1 m/s)
500 m (1600 ft)	1180 fpm (5.9 m/s)
1000 m (3300 ft)	1100 fpm (5.5 m/s)
1500 m (5000 ft)	1020 fpm (5.1 m/s)

NOTE Climb rate is measured at max continuous power (5500 RPM) of the engine with the flaps retracted at V_y and MTOM.

Climb performance may vary depending on, temperature, altitude, humidity and wing & propeller surface condition.

Cruise

Aircraft at MTOM, recommended cruise power of 5300 RPM in ISA and sea level altitude, flaps retracted:

ALPHA Trainer	ALPHA Trainer
Cruise airspeed	108 kts (200 km/h)

Best economy cruising level for the ALPHA Trainer is 6000 ft. There, cruise performance is equivalent or better than above due to IAS-TAS relation, but fuel consumption is lower. At these parameters the fuel burn is around 3.6 USgal (13.6 l) per hour. For detailed fuel consumption determination for various cruising regimes consult the Rotax 912/S Operators manual.

Effect of outside temperature

Every 10 degrees Celsius of OAT increase versus the IAS, the cruising speed at 5300 RPM decreases by 3 kts.

Descent

Reference sink rate, with flaps set to 2nd position measures 440 fpm (2.2 m/s) at 50 kts (92 km/h), power idle, airbrakes retracted.

ALPHA Trainer	ALPHA Trainer
Sink rate at 50 kts (92 km/h) full flaps	440 fpm (2.2 m/s)

The glide

The glide is defined as unpowered wings-level flight at speed providing best lift over drag ratio or minimum sink rate.

Should the engine become inoperative in flight, as a result of either intended or unintended actions, and it cannot be restarted, react as follows:

Establish wings-level flight at the speed providing best lift over drag ratio, if you desire to glide the greatest distance from a given altitude.

Establish wings-level flight at speed providing minimum sink rate, if you desire to stay airborne the longest. This may come in handy when you're forced to give way to other aircraft or if you simply need time to determine the most appropriate site to land out on.

ALPHA Trainer	ALPHA Trainer
Minimum sink speed	58 kts (108 km/h)
Minimum sink rate, flaps +15 deg	460 fpm (2.3 m/s)
Best lift/drag ratio speed	64 kts (118 km/h)
Best lift over drag ratio, flaps (0)	17:1

CAUTION: If the engine fails, especially in climb, the aircraft always loses some 30 meters (100 feet) of altitude before reaching best glide speed in wings-level unpowered flight.

Landing performance

Final approach speed should always be 55 kts (102 km/h) with full flaps. Landing runway length may also vary depending on the elevation, gross weight, touchdown velocity, wind direction and how aggressive the braking action is.

The landing roll measures 410 feet (125 meters) in the following conditions: aircraft at MTOM, airport at sea level and wind calm. Should you be flying solo, the length shortens by another 30 feet (10 meters).

Total landing distance over 50 ft / 15 m obstacle measures 1510 feet (460 m).

Landing roll increases by 10 % for every 2000 ft increase in density altitude.

Total landing distance increases by 2% for every 2000 ft increase in density altitude.

CAUTION! Minimum recommended runway length for approaches is 500 m (1640 feet) with no obstacles inside the 3 deg glide slope area and runway heading in order ensure safe flying activity. Use of shorter strips should be considered a major exception and requires a lot of skill, heavy use of slipping until the last moment before touchdown and is performed at own risk.

Crosswind takeoffs/landings

The maximum allowed crosswind speed on takeoff and landing with flaps in 2nd position is 18 kts. The runway length required increases by 10 % for every 5 kts of crosswind component.

6 Weight and balance



Introduction (6-2)

Weighing procedure (6-2)

Equipment list (6-3)

Determination of CG (6-3)

Sample CG calculation (6-4)

Introduction

This section describes the procedure for establishing the basic empty weight and moment of the airplane. Sample calculations are provided for reference.

Specific information regarding the weight and arm for this airplane as delivered from the factory can be found in the aircraft documentation folder, look for Weight and Balance Report.

WARNING! It is the responsibility of the pilot to make sure the airplane is loaded properly. Operation outside of the prescribed weight and balance limitations could result in an accident and serious or fatal injury.

Weighing procedure

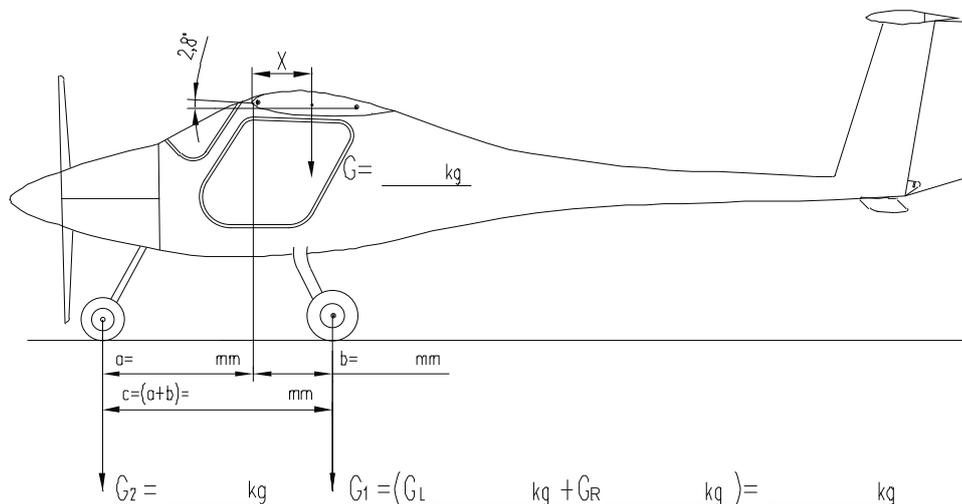
- Make sure all listed aircraft parts and appliances are installed and in position.
- Remove all other objects (e.g. tools, rugs, tie downs and other items ...).
- Empty fuel tanks except for the unusable fuel, inflate tires to recommended operating pressures.
- Fill up engine oil to the top marking.
- Retract flaps and leave control surfaces centred.
- Level the aircraft inside a closed space - use the provided airfoil template at lower side of the wing close to the wing root and make sure its straight edge is level (horizontal).
- Once levelled, read the scale readings and subtract eventual tare weight.
- Now record all readings and fill out the bottom table.

The datum is the wing's leading edge right at the fuselage root. Calculate the lever arm of CG using this formula:

$$\text{Lever arm of CG (X)} = \frac{b \times G_1 - a \times G_2}{(G_1 + G_2)}$$

Weighing form

Weighing point and symbol	Scale reading	Tare	Net
Right main wheel (GR)			
Left main wheel (GL)			
$G_1 = G_R + G_L$			
Nose wheel (G2)			
Total ($G = G_R + G_L + G_2$)			



Equipment list

The aircraft's empty weight data is unique for each and every ALPHA Trainer delivered. The owner is responsible for keeping the equipment list up to date. Fill it out according to actual status.

ALPHA Trainer

Serial number:

Registration number:

Equipment installed:

Determination of CG

	Weight (kg)	Weight's lever arm (mm)	Moment (mm x kg)	Remarks
Basic cfg. empty weight				
Baggage		930		
Instruments		-320		minus!!!
Pilots		370		
Fuel		1130		

CAUTION! Each newly installed part or appliance must be registered in the upper table. Also, new total weight and lever arm of CG values must be entered and position of CG re-determined. Furthermore, the moment must be recalculated. This is rather easy to do. First multiply the new part's weight by it's lever arm measured from the reference point (wing's leading edge). Then sum up all moments and divide the sum by the new total weight.

WARNING! The aircraft's safe center of gravity position ranges between 8.7" (220 mm) and 15.0" (382 mm) aft of datum and is not critically affected by cockpit crew weight.

WARNING! Absolute safe measure for the amount of luggage is 22 lbs (10 kg). The actual amount of luggage you can safely transport depends on the center of gravity of empty aircraft. See next pages.

Sample c.g. calculation

Guidelines

G_{total} is the total mass of empty aircraft. All calculations can be performed with aircraft empty weight and empty weight center of gravity (c.g.), as the pilots sit directly below the center of gravity and do not cause the c.g. to be shifted.

WARNING! Both pilots' weight does not influence c.g. or their influence is insignificant. However, baggage and/or fuel can influence the c.g. severely and may cause the aircraft to become uncontrollable!

Basic CG formulas and calculation

Read thoroughly. Note also that the basic c.g. at 275 mm will be used purely as an example.

First, weigh the aircraft according to the procedure described in this chapter and write down values of G₁ (sum of scale readings at main wheels) and G₂ (scale reading at tail/front wheel). Then calculate the position of c.g. in millimetres (mm) from the datum (wing's leading edge at wing root).

Use the following formula:

$$CG_{(mm)} = \frac{b \times G_1 - a \times G_2}{(G_1 + G_2)} = \frac{505 \times G_1 - 1020 \times G_2}{(G_1 + G_2)} = 275 \text{ mm}$$

where:

a is the distance from nose wheel axis to wing's leading edge,

b is the distance from main wheel axis to wing's leading edge,

Second, determine the c.g. position in percentage (%) of Mean Aerodynamic Chord (MAC) with following the formula:

$$CG_{\%MAC} = \frac{CG_{mm} \times R}{MAC} \times 100 = \frac{275 \times 40}{900} \times 100 = 26.1 \%MAC$$

where:

CG_{mm} is the position of CG in millimetres (mm),

R is the difference between wing's leading edge and MAC's leading edge (40 mm),

MAC is the Mean Aerodynamic Chord (900 mm).

Fuel and CG

The fuel shifts the CG backward, however the aircraft is factory tuned so that a full load of fuel and luggage the CG stays forward of the aft allowable limit. Similarly, when the fuel quantity is empty and there is no baggage on board, the CG is backwards of the forward allowable limit. In every case, the fuel does not shift the CG out of limits.

Baggage and CG

The amount of baggage you can carry in the baggage behind the seats is limited by the center of gravity of the empty aircraft (pilots' weight does not significantly influence c.g.) and by the MTOM.

To calculate how much the c.g. shifts because of added baggage into the solid baggage compartment or the baggage behind the seats use the following formula:

$$CG_{\text{with bags}} = \frac{(G_{\text{tot}} \times CG_{\text{mm}}) + (G_{\text{bags}} \times L_{\text{mm}})}{G_{\text{tot}} + G_{\text{bags}}} = \frac{(275 \text{ kg} \times 275 \text{ mm}) + (10 \text{ kg} \times 1160 \text{ mm})}{(275 \text{ kg} + 10 \text{ kg})} = 306 \text{ mm}$$

where:

G_{total} is the aircraft empty weight,

CG_{mm} is the position of CG of empty aircraft in millimetres (mm),

G_{bags} is the weight of the baggage,

L_{bags} is the lever arm from the datum to baggage area (1160 mm).

Again, express the new c.g. in percentage of MAC:

where:

CG_{with.bags} is the position of CG now with bags in millimetres (mm),

R is the difference between wing's leading edge and MAC's leading edge (40 mm),

MAC is the Mean Aerodynamic Chord (900 mm).

$$CG_{\text{with bags, \%MAC}} = \frac{CG_{\text{with bags}} \times R}{MAC} \times 100 = \frac{306 \times 40}{900} \times 100 = 29.6 \% \text{MAC}$$

We now have the data of c.g. of the sample aircraft with 10 kgs (22 lbs) of baggage. You can recalculate the formulas using the weights and c.g. of your empty aircraft and the planned amount of baggage for your flight.

CAUTION: The baggage weight limitations in this manual represent fool-proof limits for safe operation, even without special c.g. calculation. However, the actual baggage weight limitation is different of each individual aircraft and can be determined using the above formulas. The decision of how much baggage to carry on a flight is the pure responsibility of the pilot in command!

WARNING! Always make sure that the baggage is placed fixed inside the baggage area. Movements of baggage in-flight will cause shifts of center of gravity!

WARNING! Do not, under any circumstances attempt to fly the aircraft outside the allowable c.g. limits! Allowable c.g. range is between 8.7" and 15.0", measured from the wing's leading edge backwards.

WARNING! Maximum takeoff weight (MTOM) MUST NOT, under any circumstances, exceed 1040 lbs (472.5 kg).



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7 Aircraft & systems



Introduction (7-2)

Cockpit controls (7-4)

Instrument panel (7-4)

Undercarriage (7-6)

Seats and safety harnesses (7-6)

Pitot-static system (7-6)

Powerplant (7-7)

Fuel system (7-8)

Electrical system (7-9)

Engine cooling system (7-10)

Engine lubrication system (7-11)

Wheel brake system (7-11)

ELT system (7-12)

Electric Airbrakes (7-13)

Introduction

The Pipistrel ALPHA Trainer is a 34' 6" (10.5 m) wingspan, two-seat T-tail high-wing airplane made almost entirely of composite materials.

It has a robust, tricycle undercarriage that incorporates brake-equipped wheels, a U-shaped composite strut and a steerable nose wheel.

ALPHA Trainer features flaperons, meaning that one movable surface on each wing acts both as the flap and the aileron.

Flaps offer 3 settings: retracted 0°, extracted 15° and extracted 25°.

Full dual main flight control levers make the ALPHA Trainer ideal for initial and advanced flight training. All aileron, elevator and flap controls are connected to the cabin controls using self-fitting push-pull tubes. The rudder is controlled via cables. The elevator trim is electric.

Electrically operated airbrakes are available as an option.

All aircraft ship with H type safety belts attached to the fuselage at three mounting points. The rudder pedals can be adjusted before and also in-flight to suit your size and needs.

The fuel tank is located in the fuselage with the fuel shut-off valve located in the cockpit. The gascolator is located beneath the lower engine cover. Refuelling can be done by pouring fuel through the fuel filler neck on the fuselage or by using an electrical fuel pump via the single point refuelling valve on the bottom of firewall.

All transparent surfaces are made of 2 mm anti UV GE tinted Lexan, which was specially developed not to shatter or split on impact.

The main wheel brakes are hydraulically driven disc type and activated via a cockpit hand-lever. The hydraulic brake fluid used is DOT 4.

Cabin ventilation is achieved through special vents fitted into the doors, cabin heating and windshield defrost/demist, is provided utilizing hot air from the engine.

The propeller is a fixed pitch two-blade design.

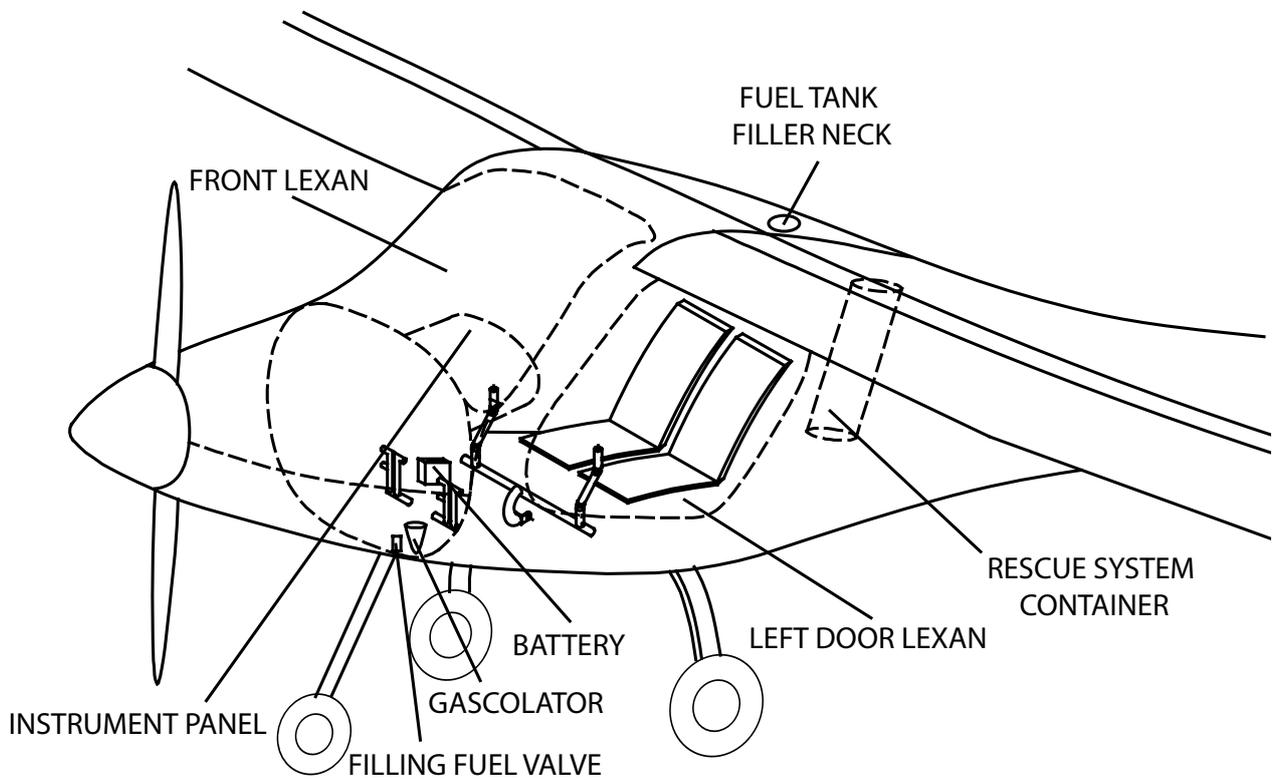
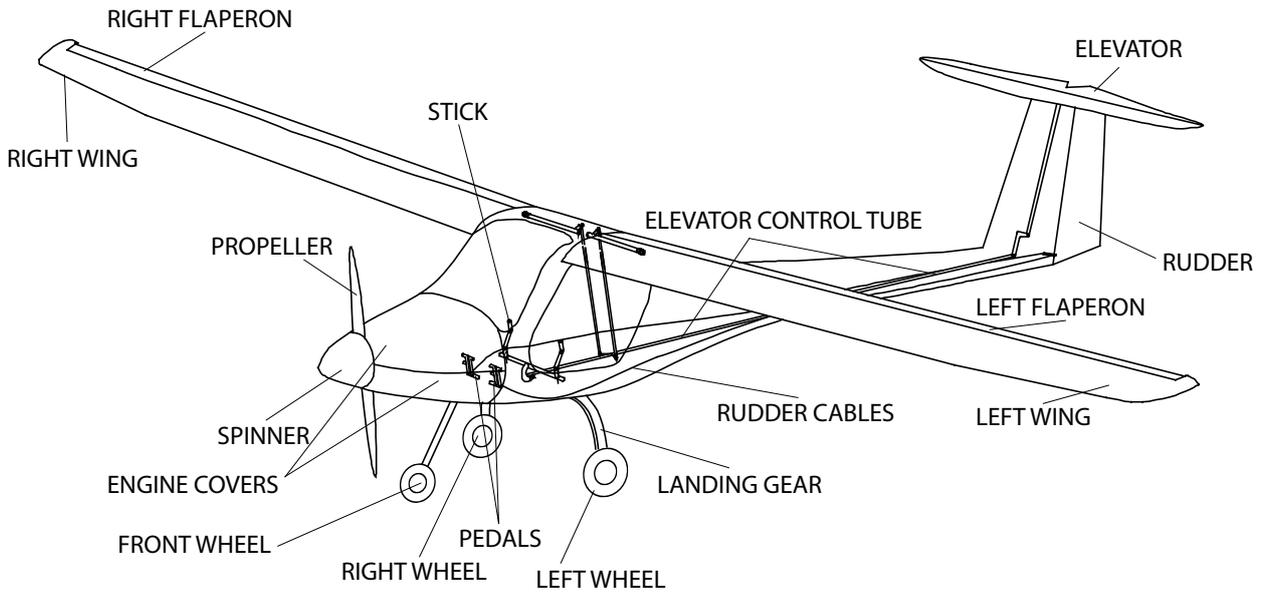
The electric circuit enables the pilot to test individual circuit items and to disconnect the battery from the circuit, should there be a distress situation.

Navigational (NAV) lights, anti collision (AC) lights and a landing (LDG) light are installed.

The firewall is reinforced with heat and noise insulation.

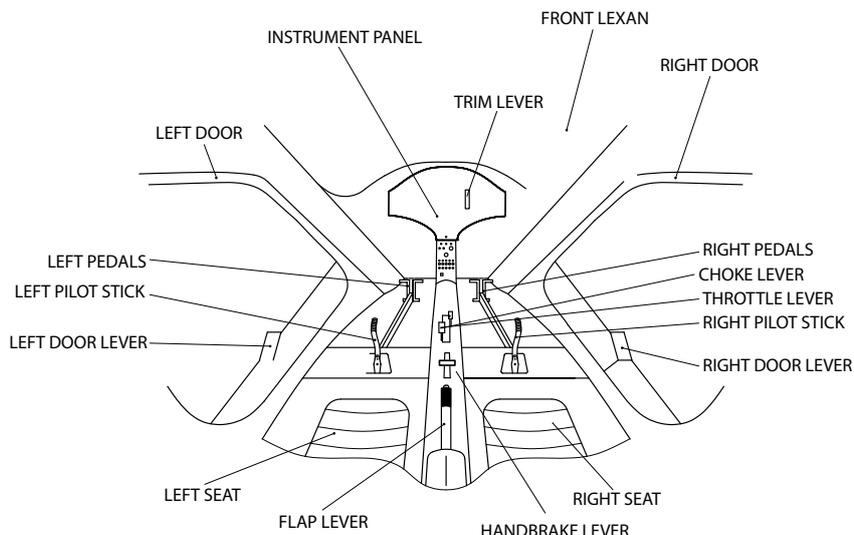
Basic instruments come installed with operational limits pre-designated.

A PRS is present and located in aft fuselage.



Cockpit controls

ALPHA Trainer cockpit levers are divided into two groups:

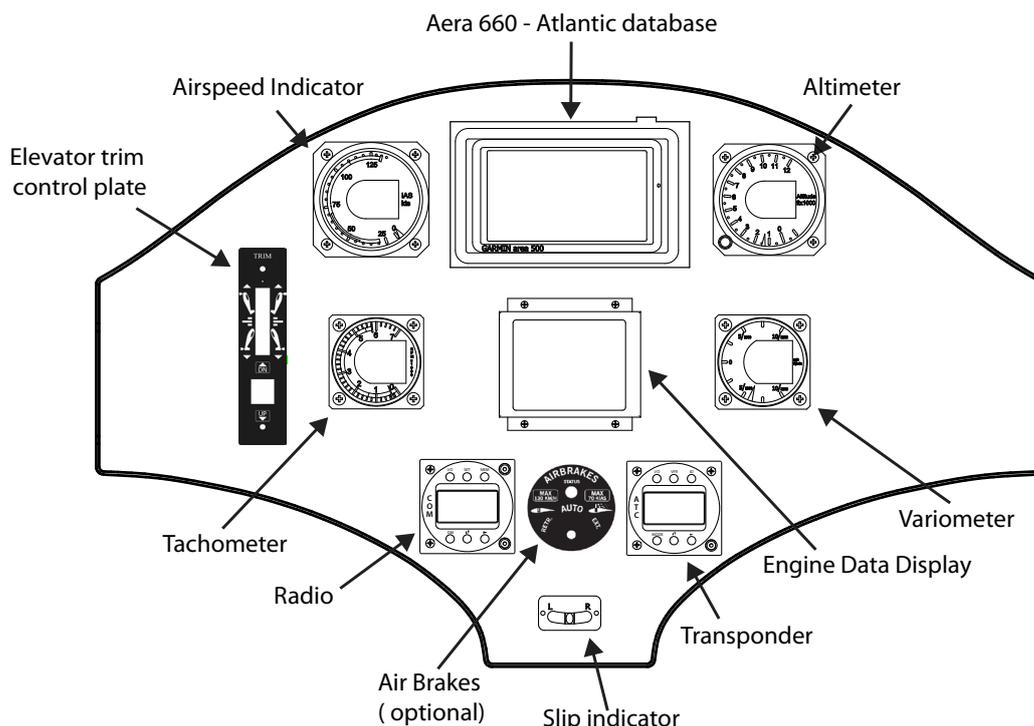


Individual control levers: pilot stick and rudder pedals

Dual control levers: throttle lever, handbrake lever, choke lever, flap lever, trim lever, battery disconnection lever/ring and emergency parachute release handle.

Instrument panel

The instrument panel is fitted with analogue gauges for: airspeed, altitude, magnetic compass, engine parameters (RPM, EGT, COOLANT TEMPERATURE, OIL TEMP, OIL PRESS, FUEL QTY gauge, Fuel pressure gauge, COM (Typically ATR833), Transponder (Typically TRT800) and a GPS (Typically Garmin AERA series). Instructions on how to use individual equipment (COM, GPS) are found in individual equipment manuals, as supplement to this POH. The gauges are round, 80 mm or 57 mm diameter, with the GPS being a touchscreen unit. Radio is a modern lightweight unit, features full intercom and dual PTT connections.



Airplane can be optionally equipped with convenience outlets, dual 2.0 USB ports are located on the right side of the instrument panel. The USB port may be used with devices that are compatible with USB 2.0 ports and can be used to power portable equipment non essential to the flight. Amperage draw through the ports should not exceed 2A.

NOTE The actual instrument configuration might vary from the one reflected above, as some optional equipment may be installed (Garmin G5, Kanardia Horis, etc.). In such case, please refer to the manufacturer OEM for details.

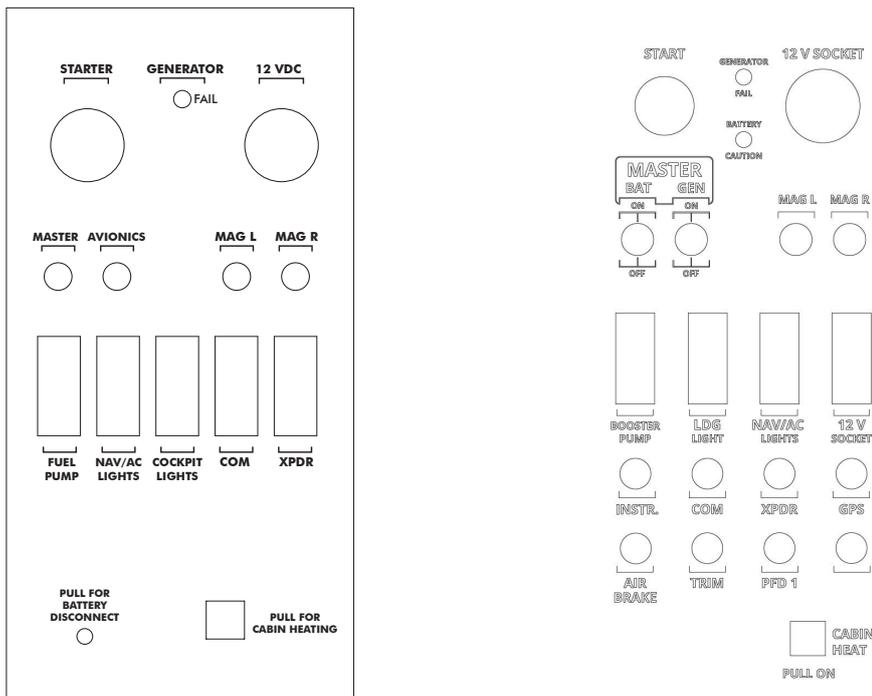
Cockpit electrical system panel:

The cockpit electrical system panel incorporates a separate master / battery switch, avionics / generator switch and engine starter push-button, separate magneto control. There are fuses located in the illuminated rectangular toggle switches, as well as wired behind the avionics / generator switch to each of the electrical avionics loads (COM, XPDR, GPS).

NOTE Procedures and descriptions in this document outline the configuration with FUEL (BOOSTER) PUMP switch installed on the switch panel, according to recommended implementation of SB-160-00-80-005 Fuel pump switch retrofit. Some older aircraft that are not compliant with any of the two service documents, may have the old configuration with the electric fuel pump connected to master / battery switch or to respective circuit breaker.

When using the cockpit electrical system panel, use the following sequence:

MOTOR START-UP		MOTOR SHUT-DOWN	
MASTER / BATTERY SWITCH	ON	OTHER SWITCHES	OFF
AVIONICS / GENERATOR SWITCH	ON	FUEL (BOOSTER) PUMP	OFF
MAG (L&R)	ON	MAG (L&R)	OFF
STARTER	ENGAGE	AVIONICS / GENERATOR SWITCH	OFF
OTHER SWITCHES	ON as desired	MASTER / BATTERY SWITCH	OFF
FUEL (BOOSTER) PUMP	ON before take-off		



NOTE These are two examples of typical switch panel of ALPHA Trainer UL first example has certain fuses installed on the main electro board. Second example of switch panel has Circuit Breakers installed on the panel itself - those are placed under the green switches on this panel.

Undercarriage

The tricycle undercarriage incorporates brake-equipped wheels, a U-shaped composite strut and a steerable nose wheel. The rudder pedals are used to steer the nose wheel. Airplane can be optionally equipped with CFRP wheel fairings.

Distance between main wheels:	63 inch (1.6 m)
Distance between main and nose wheel:	60 inch (1.52 m)
Tire, 8 ply:	4,00" x 6" (main wh.), 4,00" x 4" (nose wh.)
Tire pressure:	24 psi - 28 psi (main wh.), 18 psi (nose wh.)
Brakes:	disk type, driven by brake lever centrally located in cockpit
Brake fluid:	DOT 4

The parking brake function is applied using a lock-latch on the handbrake lever. To apply the parking brake, pull handbrake lever firmly, hold it engage and use the lock-latch to activate parking brake function. To release, simply release the lock-latch, and push handbrake lever to full forward position.

Seats and safety harnesses

The seats have no stiff internal structure and can therefore be folded forward easily for access to aft fuselage. The seat has one position, whereas the pedals are adjustable. Custom made seats are available for ordering. All ALPHA Trainer ship with H type safety harness attached to the fuselage at three mounting points.

Pitot-static system

The pitot tube is attached to the bottom side of the starboard wing. The pitot lines run through the inside of the wing all the way to the instrument panel.

Power plant

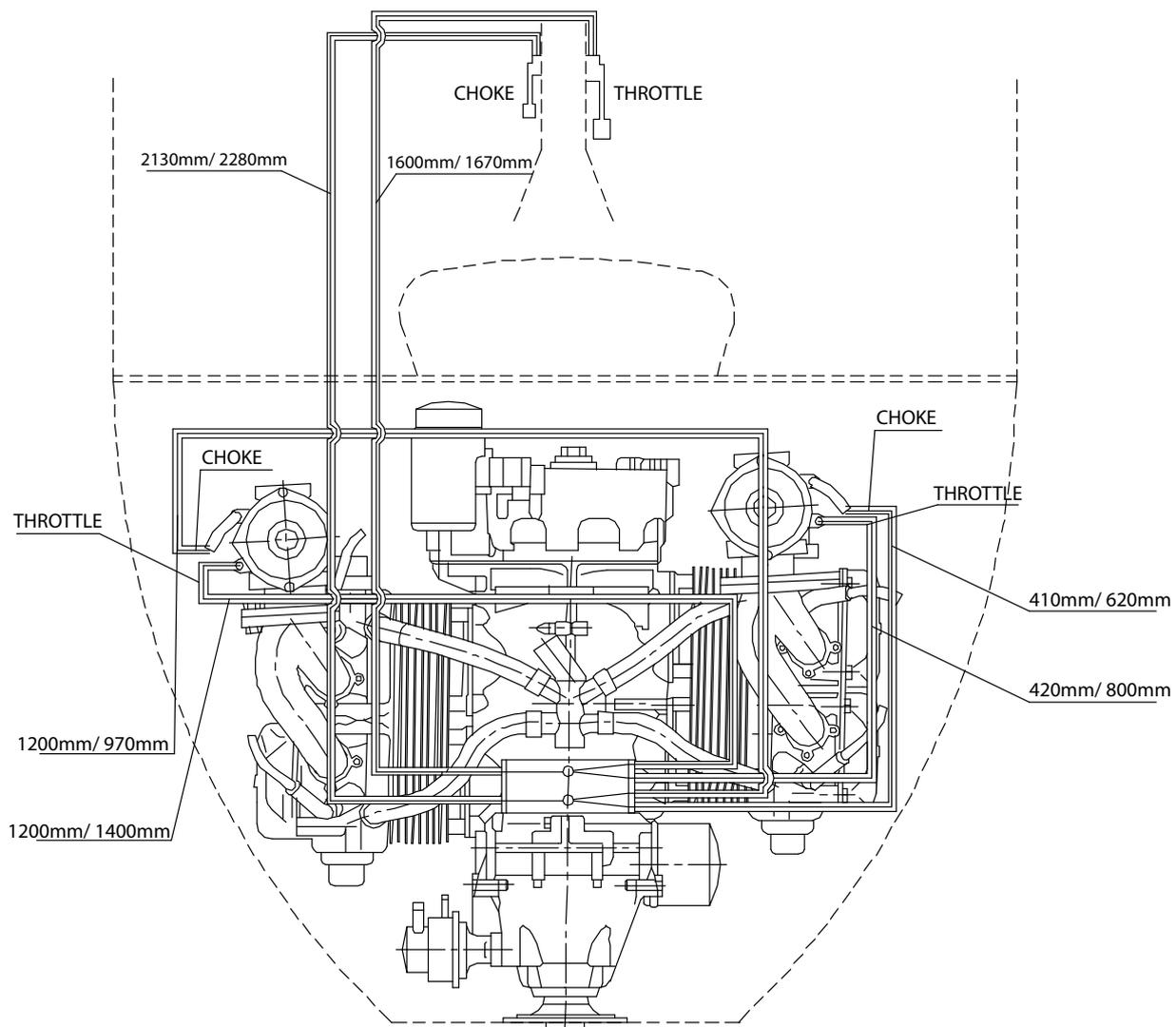
ALPHA Trainer is equipped with Rotax 912 (80 HP) engine.

Engine description:

Engine:	ROTAX 912 80 HP (4-stroke boxer, four cylinders, 1211 cm³) twin carburetted - dual electronic ignition
Cooling:	crank case air-cooled, cylinder heads water-cooled - own radiator and pump, other moving parts oilcooled - own radiator and pump
Lubrication:	centrally oiled - own oil pump and radiator
Reduction gearbox:	integrated
Reduction ratio:	1 : 2.27
El. generator output power:	250 W at 5500 RPM
Starter:	electric
Engine power:	80 HP at 5800 RPM
Battery:	12 V, 11 Ah

All metal cables used are fire resistant, kept inside bowdens i.e. self-lubricating flexible tubes.

Schematic of throttle and choke control



Propeller type:

Pipistrel FP02-80
(for Rotax 912 80 HP):

two blade, fixed pitch propeller, diameter 65" (1660 mm)

Fuel system

Description:	vented fuselage fuel tank refuelling opening on top/side of fuselage
Fuel shutoff valve:	one, located in cockpit
Gascolator:	filter equipped with drain valve
Fuel capacity:	13.2 USgal (50 liters)
Unusable fuel:	0.5 USgal (2 liters)
Fuel filter:	metal, inside the gascolator
Fuel (Booster) pump:	electrical: activated by a dedicated switch (or linked to master / battery switch or dedicated CB - see switch panel description)

All fuel hoses are protected with certified glass-silicon rubber. ALPHA Trainer's fuel system features fuel return circuit into the fuel tank.

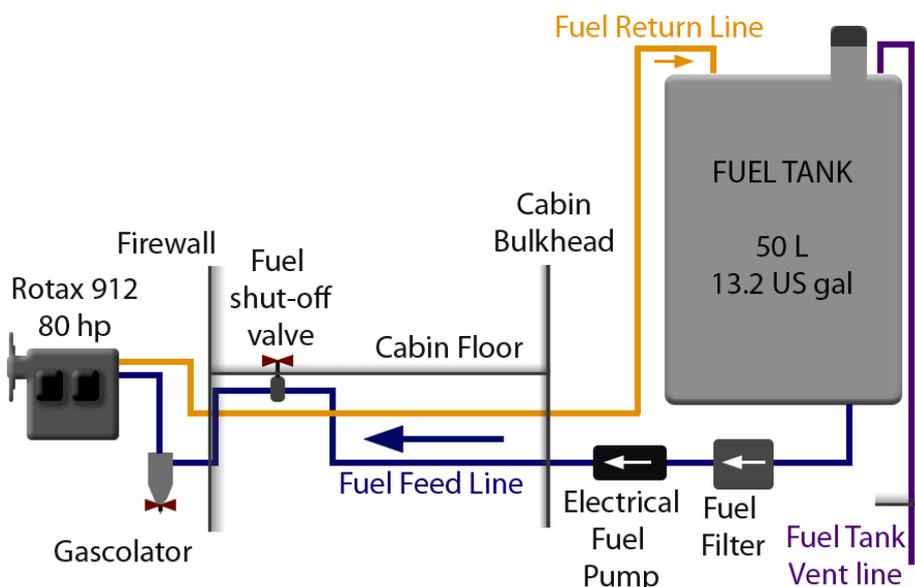
There is an electric - gauge style fuel quantity indication for the pilot.

CAUTION! Flying in sideslip or turbulence may result in imprecise fuel quantity indication. Pilot caution is advised.

NOTE The actual fuel consumption/flow and fuel quantity remaining are calculations based on engine rpm and manifold pressure. These settings are to be used as a guide only and are NOT to be relied upon for the safe operation of the aircraft. If in doubt confirm the fuel quantity remaining before flight.

Draining of water and/or particles is carried out by draining the contents of the gascolator, installed below the bottom engine cowl and reachable through a dedicated placarded opening. Unscrew the discharge valve and drain at least 1 cup of fuel in a transparent canister, verify for water/particle contamination. Always fasten the drain valve before flight!

When using the single point fuel valve, found below the draining opening (placarded), make sure you have closed it before flight. The single point fuel valve can either be used for fuelling the aircraft by using a pump and container, or for discharging all of the fuel on board before disassembling the aircraft.



Electrical system

Description:	dual electronic ignition. Standard, 12 V circuit charges the battery and provides power to all appliances and instruments.
Master / battery switch:	toggle switch type
Avionics / generator switch:	toggle switch type, powers all electrical avionics.
Magneto switches:	separate Left (MAG L), Right (MAG R)
Other switches:	fused and equipped with control lights (switch illumination)
Battery:	12 V, 11 Ah
Measured power consumption of some electrical loads:	Nav/Strobe lights: 1 (steady) - 2 (peak) A , Cockpit light: 0.5 A, Radio & Transponder, EFIS, Please consult item's operating manual

The electrical system uses simplified architecture. Characteristic are separate magneto toggle switches and a switch-type master / battery switch. Avionics / generator switch is separate. There are individual fused rocker switches used to control individual electrical loads (lights, etc.), apart from electrical avionics.

Alternator (AC) is integral to the engine and provides up to 170 Watts at 5500 RPM. It is connected to the rectifier with an output of 14 V (DC).

Correct sequence of use for the new type of cockpit electrical panel is are as follows:

Engine start-up: MASTER / BATTERY ON - AVIONICS / GENERATOR ON - START ENGINE.

Engine shut-down: AVIONICS / GENERATOR OFF - ENGINE OFF - MASTER / BATTERY OFF.

The electrical system itself includes three solenoids, one activated by the master / battery switch, the second activated by the avionics / generator switch and the starter engage. All electrical loads, apart from the 12 V socket and the Pitot heat, are connected to the avionics bus via push-pull circuit breakers (after implementation of *SB-160-00-80-005 Fuel pump switch retrofit*, 12V socket is connected to respective CB/Fuse too). For loads, which are engaged and disengaged more often, fused rocker switches are used (NAV lights, LDG light, etc.) All other loads (e.g. avionics) receive power as soon as the avionics / generator switch is ON. The avionics / generator switch has no function when the master / battery switch is OFF. The starter engage button is also disabled when the master / battery switch is OFF.

Battery disconnection system (optional)

On the ALPHA Trainer, the main battery can be disconnected from the circuit.

There are two handles in the cockpit used to operate the battery disconnection, the battery disconnection lever and the battery disconnection ring. The battery disconnection lever, which is a red flag-type lever is found on the firewall above the main battery on the left-hand side of the cockpit. This lever has an attached wire which leads to the battery disconnection ring on the instrument panel's switch column.

To disconnect the battery from the circuit, simply pull the battery disconnection ring on the instrument panel's switch column.

To reconnect the battery back to the circuit, use the flag-type lever on the firewall.

Deflect the lever so that its flag end points towards the firewall. Having done this correctly, you will feel the flag-lever lock into position.

Battery reconnection can be done in-flight as well (e.g. following a successfully rectified emergency situation) but only from the left-hand seat, since you cannot reach the flag-lever from the right-hand side of the cockpit.

Engine cooling system

Rotax 912 (80 HP) cooling system

The Rotax 912 engine's cylinders are air-cooled, the cylinder heads are liquid cooled. The cooling-air intake is located on the right-hand bottom part of the engine cover.

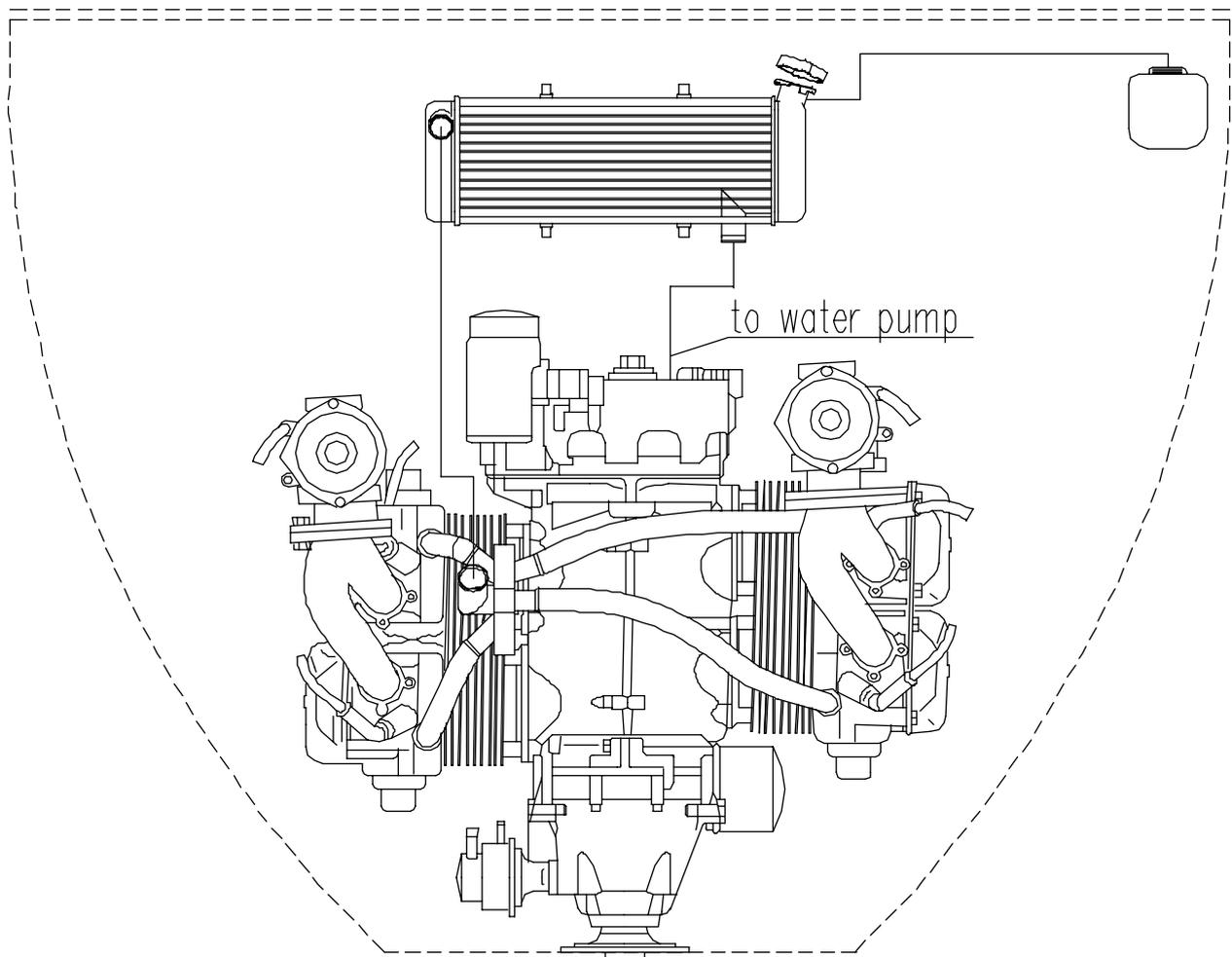
Cylinder heads are liquid cooled. The water pump forces water through the radiator. The engine does not feature a thermostat valve. The overflow tank fluid level must always be inside the designated limits!

The engine does not feature a cooling fan, therefore cooling is entirely dependent on moving air currents and airspeed.

CAUTION! You are strongly discouraged from leaving the engine running at idle power when on ground for extended periods of time.

Consult Rotax Service Information SI-912-016 for the list of applicable and approved coolant media.

Schematic of engine cooling system

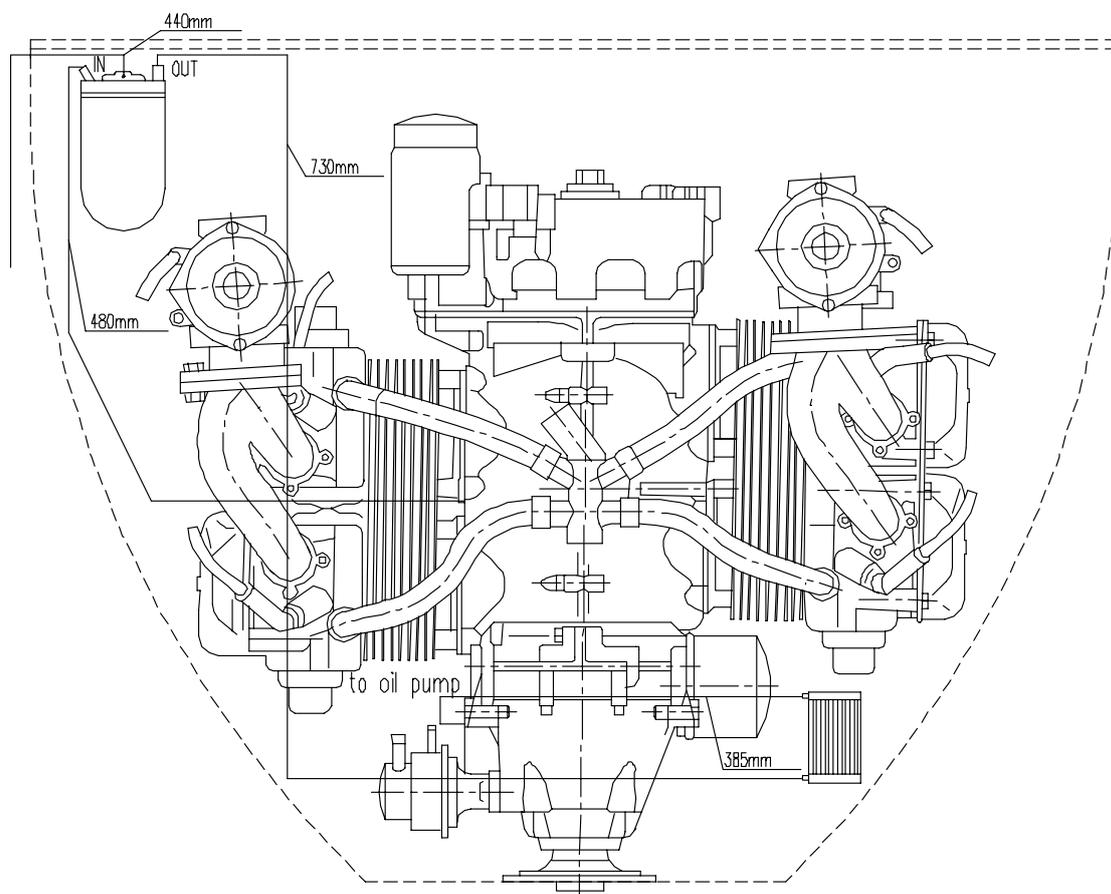


Engine lubrication system

Rotax 912 is a four-stroke engine, equipped with a dry sump and lubricated centrally with use of its own oil pump. All the oil needed is located inside a separate canister. When the engine is running, the oil cools itself passing through a radiator. Oil quantity can be checked visually with a oil dip stick. Make sure the oil quantity is within limits at all times. Airplane can be optionally equipped with oil thermostat. Once the oil temperature reaches its limit, the thermostat opens, allowing the oil to flow through the coolers.

CAUTION! Oil temperature, pressure and quality is strictly defined and must not, under any circumstances, vary from its safe values.

Schematic of engine lubrication system



Wheel brake system

The wheel brakes are disc, hydraulic type, actuated together by pulling on the common handbrake lever.

The hydraulic brake fluid used is DOT 4.

The parking brake function is engaged using a lock-latch on the handbrake lever. To apply the parking brake, pull handbrake lever firmly, hold it engage and use the lock-latch to activate parking brake function. To release, simply release the lock-latch, and push handbrake lever to full forward position.

ELT (Emergency Locator Transmitter)

The ALPHA Trainer can be equipped with an Emergency Locator Transmitter (ELT). The device is composed

by the ELT transmitter, installed behind the right seat, a remote switch mounted on the instrument panel, connecting wires and antenna. In the event of an aircraft accident, these devices are designed to transmit a distress signal on different frequencies (121.5 and 406 MHz). The ELT is activated either automatically, once a certain force threshold is reached (e.g. in case of incident) or manually, by setting the remote switch, installed on the instrument panel, to ON. Two ELT models can be found on the aircraft: Kannad 406 AF or Artex 345.

For correct operation of the system, following switches position must be ensured:

	Artex 345		Kannad 406 AF	
	ELT (box)	Remote switch	ELT (box)	Remote switch
Flight position (normal):	ARM/OFF	ARM/OFF	ARM	ARMED
Manual activation during emergency landing:	use remote switch	ON	use remote switch	ON
Manual activation (after emergency landing):	ON	ON	ON	ON
End transmission:	ON>ARM/OFF	ON>ARM/OFF	OFF	RESET/TEST

Electric airbrake system

The optional system consists of 2 individually actuated airbrake assemblies with accompanying servo motors, a drive-module, flap position sensor and instrument panel knob/LED assembly.

All connections between the modules are electric. Make sure that the electrical connectors to the wings are secured when assembling/disassembling the aircraft.

The system has three (3) modes:

Automatic (AUTO) - airbrakes are extended when flaps are extended to 25°.

Manual Retracted (RETR.) - airbrakes are retracted, regardless of the flap position.

Manual Extended (EXT.) - airbrakes are extended, regardless of the flap position.

The airbrakes should be calibrated before each flight, as described in the cockpit pre-flight inspection. During this procedure the system determines and records the position/movement range of the airbrakes. If the calibration procedure is not performed, the system will use the previously stored values.

CAUTION: Not carrying out the calibration procedure can lead to deviations in airbrake position and excessive loads in the system.

The calibration procedure is activated by turning the airbrakes control knob to EXTENDED within 5 seconds after the MASTER / BATTERY switch has been engaged. The procedure also begins if the knob is already in EXT position when the MASTER / BATTERY switch is engaged. Once the procedure begins, the airbrakes open fully for three seconds, then close for three seconds and finally return to the EXTENDED position, as indicated by the control knob.

The control panel features a STATUS LED indicator.

See SPOH-161-00-41-050 - Electric airbrakes POH supplement for additional information.

NORMAL Blinking codes:

Airbrakes Extended >> LED On (solid orange)

Airbrakes Retracted >> LED Off

ERROR Blinking codes:

1) **Two blinks per second:** airbrake/s jammed due to mechanical or electrical problem during normal operation.

The other airbrake will move to the position of the jammed one in order to reach a symmetrical position.

Try to eliminate the problem by extracting/retracting the airbrakes, using the control knob.

If the problem persists, try recalibrating the airbrakes after landing.

2) **Fast blinking (10 Hz):** calibration procedure error due to mechanical jamming or electrical problem.



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8 Handling and maintenance



Special inspections (8-2)

Draining and refueling (8-2)

**Connecting auxiliary power
supplies (8-3)**

Tie down (8-4)

Storage (8-4)

Cleaning (8-4)

**Keeping your aircraft in
perfect shape (8-5)**

Special inspections

After having exceeded VNE or landed in a rough manner:

Check the undercarriage, fuselage & wing surfaces and main spars for abnormalities. It is highly recommended to have the aircraft verified for airworthiness by authorized service personnel.

Clicking noise overhead

The wings are factory fitted to the fuselage to make a tight fit at approximately 80 F (27°C). When exposed to low temperatures, materials shrink. Therefore, flying in the winter or in cold temperatures, you may encounter "click-clack" like noises above your head. The remedy for this unpleasant noises is to add washers, typically of 0,5 mm thickness in-between wing and fuselage. Washers must be added both at rear and front bushings at one side of the fuselage only!

WARNING! It is mandatory to consult the manufacturer or authorised service personnel before applying washers!

Draining and refuelling

Whenever draining or refuelling make sure master / battery switch is set to OFF.

Draining the fuel system

The gascolator is located beneath the bottom engine cover on the left hand side of the fuselage. To drain the fuel system, open the drain valve on the gascolator. Drain approximately 1/2 cup of fuel. Try to prevent ground pollution by collecting the fuel with a canister. To close the valve simply turn it in the opposite direction. **Do not use force or special tools!**

CAUTION! Always drain the fuel system before you have moved the aircraft from a standstill to prevent mixing of the fuel and eventual water or particles.

Refuelling

CAUTION! Before refueling it is necessary to ground the aircraft - connect ground wire from re-fuel nozzle to airplane exhaust, from airplane exhaust to fuel truck or cart, and from fuel truck or cart to a suitable earth ground.

Refuelling can be done by pouring fuel through the fuel tank opening on top/side of the fuselage or by using the single point fuelling valve on the lower firewall.

Connecting auxiliary power supplies

Should you be unable to start the engine due to a weak battery, auxiliary power supplies can be connected to help starting the engine. The below photos are for illustration only and certain elements shown do not depict the actual situation on the aircraft.

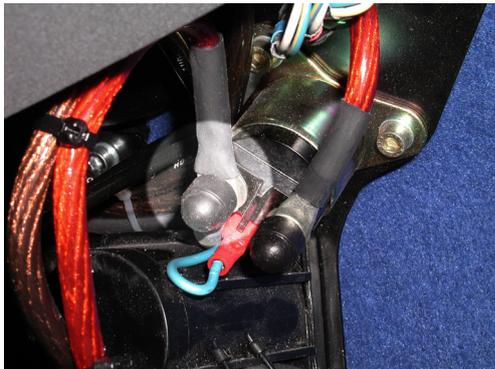
Battery's & Relay's location



Battery (black) & Relay (top-right)



Top-left nipple (c. positive (+) wire here)



Exhaust (connect negative (-) wire here)



To connect an auxiliary power supply use battery connector cables with clamps at either ends. Connect the negative (-) wire to aircraft's exhaust (sticking out below the engine cowling). The positive (+) wire leads inside the cockpit to the relay mounted top-right of the aircraft's battery on the firewall. This relay has 3 nipples; the positive (+) wire must be connected to the upper-left nipple, the only one to which 2 cables are connected to. After you have connected the wires correctly, start the engine normally by pressing the starter button in the cockpit.

WARNING! The pilot must be in cockpit when starting the engine. The person who will disconnect the cables after the engine has started must be aware of the danger of spinning propeller nearby.

Tie down

Point the aircraft into the wind and retract flaps fully. Chock all three wheels. Put an extra rope around the tail-cone and into the slot between the propeller and the spinner. When using rope of a non-synthetic material, leave sufficient slack to avoid damage to the aircraft, should the ropes contract. To tie down the tail, tie a rope through the tail skid and secure it to the ground. At the end, cover the pitot tube with a protection cover. Airplane can be optionally equipped with wing tie down points for additional securing. In this case attach the tie-down rings to the wings, secure the tie-down ropes to the tie-down rings on the wing at approximately 45 degree angle to the ground. Anchor points for wing tie-downs should not be more than 5 m apart to prevent wing tie-down rings damage in heavy winds.

Storage

The aircraft should ideally be stored in a hangar. For increased in-hangar manoeuvrability use of original push-cart is recommended.

The parachute rescue system is installed in your aircraft, so make sure the activation handle safety pin is inserted every time you leave the aircraft.

Also, disconnect the battery from the circuit to prevent battery self-discharge (pull battery disconnection ring on the instrument panel's switch column) during storage period.

Cleaning

Use fresh water and a soft piece of cloth to clean the aircraft's exterior. If you are unable to remove certain spots, consider using mild detergents. Afterwards, rinse the entire surface thoroughly.

Lexan transparent surfaces are protected by an anti-scratch layer on the outside. Always use fresh water only to clean the transparent surfaces, not to damage these protection layers and coatings.

To protect the aircraft's surface (excluding transparent surfaces) from the environmental contaminants, use high-quality car wax.

The interior is to be cleaned with a vacuum cleaner.

Keeping your aircraft in perfect shape

Precautions

- 1) Eliminate the use of ALL aggressive cleaning solutions and organic solvents, also the window cleaning spray, benzene, acetone, aggressive shampoos etc.
- 2) If you must use an organic solvent (acetone) on small areas remove certain glue leftovers or similar, the surface in question MUST be polished thereafter. The only section where polishing should be avoided is the edge on the wing where the sealing gasket is applied.
- 3) When flying in regions with a lot of bugs in the air, you should protect the leading edges of the airframe before flight (propeller, wings, tail) with Antistatic furniture spray cleaner: "Pronto (transparent), manufacturer: Johnson Wax (or anything equivalent) – Worldwide", approximate price is only \$3 USD / €3 EUR for a 300 ml spray bottle. Using such spray, do not apply it directly onto the wing but into a soft cloth instead (old T-shirts are best).
- 4) After having finished with flight activity for the day, clean the leading edges of the airframe as soon as possible with a lot of water and a drying towel (chamois, artificial leather skin). This will be very easy to do if you applied a coat of Pronto before flight.

Detailed handling (Airframe cleaning instructions)

Every-day care after flight

Bugs, which represent the most of the dirt to be found on the airframe, are to be removed with clean water and a soft cloth (can be also drying towel, chamois, artificial leather skin). To save time, soak all the leading edges of the airframe first. Make sure to wipe ALL of the aircraft's surface until it is completely dry at the end.

Clean the propeller and the areas with eventual greasy spots separately using a mild car shampoo with a wax.

CAUTION! Do not, under any circumstances attempt to use aggressive cleaning solutions, as you will severely damage the lacquer, which is the only protective layer before the structural laminate.

When using the aircraft in difficult atmospheric conditions (intense sunshine, dusty winds, coastline, acid rains etc.) make sure to clean the outer surface more thoroughly.

If you notice you cannot remove the bug-spots from the leading edges of the aircraft, this means the lacquer is not protected any more, therefore it is necessary to polish these surfaces.

CAUTION! Do not, under any circumstances attempt to remove such bug-spots with abrasive sponges and/or rough polishing pastes.

Periodical cleaning of all outer surfaces with car shampoo

Clean as you would clean your car starting at the top and working your way downwards using a soft sponge. Be careful not to use a sponge that was contaminated with particles e.g. mud, fine sand) not to grind the surface. While cleaning, soak the surface and the sponge many, many times. Use a separate sponge to clean the bottom fuselage, as is it usually more greasy than the rest of the airframe. When pouring water over the airframe, be careful not to direct it over the fuel reservoir caps, wing-fuselage joining section, parachute rescue system straps and cover, pitot tube, tail static probe and engine covers.

Always rinse the shampooed surfaces again before they become dry! Thereafter, wipe the whole of the aircraft dry using a drying towel, chamois or artificial leather skin.

Also, clean the Mylar seals on the wing and tail control surfaces. Lift the seals gently and insert ONE layer of cloth underneath, then move along the whole span of the seal. Ultimately, you may wish to apply Teflon grease (in spray) over the area where the seal touch the control surfaces.

Polishing by hand

Use only the highest quality polishing compounds WITHOUT abrasive grain, such as Sonax Extreme or similar. Start polishing on a clean, dry and cool surface, never in the sunshine!

Machine polishing requires more skills and has its own particularities, therefore it is recommended to leave it to a professional.

Cleaning the Lexan transparent surfaces

It is most important to use really clean water (no cleaning solutions are necessary) and a really clean drying towel (always use a separate towel ONLY for the transparent surfaces). Should the transparent surfaces be dusty, remove the dust first by pouring water (not spraying!) and gliding your hand over the surface. Using the drying towel, simply glide it over the surface, then squeeze it and soak it before touching the surface again. If there are bugs on the windshield, soak them with plenty of water first, so less wiping is necessary. Ultimately, dry the whole surface and apply JT Plexus Spray (\$10 USD / €10 EUR per spray) or at least Pronto antistatic (transparent) spray and wipe clean with a separate soft cotton cloth.

9 Appendix



**Parachute rescue system:
use, handling and
maintenance (9-2)**

Training supplement (9-4)

Conversion tables (9-8)

**Pre-flight check-up
pictures (9-4)**

Parachute rescue system: use, handling and maintenance

System description

The PRS- parachute rescue system provides you with a chance to rescue yourself from an unexpected situation.

The system is placed inside a durable cylinder mounted on the right hand side of the baggage compartment. Inside this cylinder is the parachute which stored inside a deployment bag with a rocket engine underneath.

This brand new design deploys a canopy that is not gradually drawn from the container, exposed to distortion by air currents, but it is safely open after 0,4 to 0,7 seconds in distance of 50-60 ft above the aircraft. It is carried there in a special deployment bag, which decreases the risk of aircraft debris fouling the canopy.

The parachute rescue system is activated manually, by pulling the activation handle mounted on the back wall above. After being fired, the man canopy is open and fully inflated in about 3.2 seconds.

WARNING! Activation handle safety pin should be inserted when the aircraft is parked or stored in a hangar to prevent accidental deployment. However, the instant pilot boards the aircraft, safety pin **MUST** be removed!

Use of parachute rescue system

Typical situations for use of the parachute rescue system are:

- structural failure
- mid-air collision
- loss of control over aircraft
- engine failure over hostile terrain
- pilot incapacitation (incl. heart attack, stroke, temp. blindness, disorientation...)

Prior to firing the system, provided time allows:

- shut down the engine and set master / battery switch to OFF (key in full left position)
- shut the fuel valve
- fasten safety harnesses tightly
- protect your face and body.

To deploy the parachute jerk the activation handle hard to a length of at least 15" towards the instrument panel.

Once you have pulled the handle and the rocket is deployed, it will be about two seconds before you feel the impact produced by two forces. The first force is produced by stretching of all the system. The second force follows after the inflation of the canopy from opening impact and it will seem to you that the aircraft is pulled backwards briefly. The airspeed is reduced instantly and the aircraft now starts to descent underneath the canopy.

As a pilot you should know that the phase following parachute deployment may be a great unknown and a great adventure for the crew. You will be getting into a situation for the first time, where a proper landing and the determination of the landing site are out of your control.

CAUTION! Should you end up in power lines (carrying electrical current), DO NOT under any circumstances touch any metal parts inside or outside the cockpit. This also applies to anyone attempting to help or rescue you. Be aware that anyone touching any part of the aircraft while standing on the ground will probably suffer mayor injury or die of electrocution. Therefore, you are strongly encouraged to confine your movements until qualified rescue personnel arrives at the site to assist you.

After the parachute rescue system has been used or if you suspect any possible damage to the system, do not hesitate and immediately contact the manufacturer!

Handling and maintenance

Prior to every flight all visible parts of the system must be checked for proper condition. Special attention should be paid to corrosion on the activation handle inside the cockpit. Also, main fastening straps on the outside of the fuselage must be undamaged at all times.

Furthermore, neither system, nor any of its parts should be exposed to moisture, vibration and UV radiation for long periods of time to ensure proper system operation and life.

All major repairs and damage repairs MUST be done by the manufacturer or authorized service personnel.

For all details concerning the PRS rescue system, please see the "PRS - Parachute Rescue System Manual for Assembly and Use".

Training/Familiarisation Supplement

This chapter has been written to assist owners/pilots/instructors of ALPHA Trainer on their quest to learn how to safely and efficiently fly this aircraft in addition to the information already assembled in the rest of this POH. This section will cover most operations the aircraft offers in an order established in section Normal procedures and recommended speeds. Please consider what follows as an add-on to that chapter.

Engine start-up

First and foremost **make sure you have sufficient fuel quantity on board** for the desired length of flight. If you are not completely confident there is enough, step out of the aircraft and add more fuel into the tanks. There is an old aviators' saying: *"The only time you have too much fuel is when you are on fire."*

When engaging engine starter, wheel brakes MUST be engaged. To keep your propeller in perfect shape, avoid starting up on areas where there are small stones on the ground. Those little stones can easily be picked up by the propeller causing damage to the blades.

Warming up must be conducted below 2500 RPM. When reaching safe operational engine temperatures, verify maximum engine ground RPM. **Hold the stick back completely and slowly(!) add throttle to full power, then verify RPM.**

Taxi

Taxiing with the ALPHA Trainer is rather simple considering the steerable nose wheel. It is recommended you **taxi slowly**, up to 10 km/s (5 kts), while holding the stick back fully to ease the pressure of the nose wheel.

During taxiing monitor engine temperatures. Due to low airflow around the radiators the CHT and Oil temperature will rise during long taxi periods. If you are holding position, do not leave throttle at idle. It is better you have some 2500 RPM as this will provide some airflow from the propeller to the radiators and the temperatures will not rise so quickly. **Should you see engine temperatures exceed safe operational values, shut off the engine, point the aircraft's nose into the wind and wait for the temperatures to reduce.**

Take off and initial climb

Having checked and set all engine and aircraft parameters, you should be ready for take off by now. **Reverify fuel valve is open. Trim lever should be in the middle.**

Start the take-off roll gradually. Keep adding throttle smoothly and slowly to full power. There are two reasons for this. First, you change flight stage from zero movement to acceleration slowly; this provides you with time to react to conditions. Second, especially if taking-off from a gravel runway, this method of adding full throttle will prevent the little stones on the runway to damage the propeller. Extremely short runways are an exception. There you should line up the aircraft, set flaps to 2nd stage (in this case set airbrakes to RETR), step on the brakes, apply full power and release the brakes.

As you start to move, pull the stick 1/3 of elevator's deflection backwards to ease the pressure on the nose wheel and lift it off the runway slightly. **Do not use full back deflection as this will cause the aircraft's tail to touch the ground.**

When the nose wheel has lifted off the ground, there is nothing else but to hold the same pitch attitude and the aircraft will become airborne. Crosswind take-offs, depending on wind strength, require a little bit of aileron deflection into the wind. **Remember, wings must stay level throughout ground-roll, rotation and initial climb!**

Having lifted off the ground, gently push the stick forward just a bit to accelerate. At some 110 km/h (60 kts) set flaps to 1st stage, at 130 km/h (70 kts) set them to neutral.

Climb

A comfortable setting for climb is flaps in zero/neutral position, speed of 76 kts (140 km/h) at or slightly below 5500 RPM. In summer time or **when outside temperature exceeds 85 F(30°C)** you should **consider climbing at some 85 kts (160 km/h)** to provide more airflow to the engine radiators. Trim the aircraft for comfortable stick forces.

Cruise

Make sure flaps are retracted. A comfortable cruise setting is 5300 engine RPM.

Cruising fast, do not kick-in rudder for turns! Above 85 kts (160 km/h) the rudder becomes almost insignificant in comparison to aileron deflections when it comes to making a turn. **Cruising fast, it is important to fly coordinated (ball in the middle) as this increases efficiency and decreases side-pressure onto vertical tail surfaces. Also, pay attention to turbulence. If you hit turbulence at speeds greater than VB, reduce power immediately and pull up the nose to reduce speed.**

If flying a traffic pattern, set engine power so that airspeed does not exceed 150 km/h (80 kts).

Descent

Descending with the Alpha Trainer is the stage of flight where the most care should be taken. The aircraft is aerodynamically clean and builds up speed very fast.

Start the descent by reducing throttle and keep your speed below VB.

During initial descent it is recommended you trim for a 10 kts lower speed than the one you decided to descent at. Do this for safety. In case you hit turbulence simply release forward pressure on the stick and the aircraft will slow down.

Also, keep in mind you need to begin your descent quite some time before destination. A comfortable rate of descent is 500 fpm (2.5 m/s). So it takes you 2 minutes for a 1000 ft (300 m) drop. At 105 kts (200 km/h) this means 3.6 NM for each 1000 ft drop.

Entering the traffic pattern the aircraft must slow down. In order to do this, hold your altitude and reduce throttle to idle. Gradually slow down to below 80 kts (150 km/h), then set proper engine RPM to maintain speed of 70 kts (130 km/h). Trim the aircraft for comfortable stick forces.

Before turning to base-leg, reduce power to idle and set flaps to 1st stage at 70 kts (130 km/h). Once out of the turn, reduce speed towards 60 kts (110 km/h). Power remains idle from the point of turning base all the way to touch-down. If you plan your approach this way, you will always be on the safe side - even if your engine fails, you will still be able to safely reach the runway!

Turn to final at 55 kts (100 km/h). When in runway heading, set flaps to 2nd stage. Use the throttle to obtain the desired descent path (if applicable).

Roundout (Flare) and touchdown

Your speed should be a constant 55 kts (100 km/h) throughout the final with the descent path constant as well. At a height of 10 meters (25 feet) start a gentle flare and approach the **aircraft must touch down with the main (back) wheels first**, so that you will not bounce on the runway. After touchdown, operate the rudder pedals if necessary to maintain runway heading and try to have the nose wheel off the ground for as long as possible. When the nose wheel is to touch the ground, rudder pedals **MUST** be exactly in the middle not to cause damage to the steering mechanism. **While braking, hold the stick back fully!** Once you have come to a standstill, retract flaps all the way to normal 0° position (handle full down) and retract the airbrakes.

Should you bounce off the runway after touch-down, do not, under any circumstances, push stick forward. Bouncing tends to reduce by itself anyhow.

Crosswind landings, depending on the wind speed, require some sort of drift correction. Most efficient is the low-wing method, where you are to lower the wing into the wind slightly and maintain course by applying appropriate rudder deflection. You can also try the crab method.

Crosswind landings on paved runways (asphalt, concrete, tarmac...)

In this case, special attention must be paid to straightening the aircraft before touchdown in order not to damage the undercarriage because of increased surface grip on impact.

Should the crosswind component be strong (8 kts and over), it is recommended to gently flare in such a manner, that one of the main wheels touches-down an instant before the other (e.g. if there is crosswind from your left, the left wheel should touch down just before the right wheel does). This way the undercarriage almost cannot be damaged due to side forces on cross-wind landings.

Landing in strong turbulence and/or gusty winds

First of all airspeed must be increased for half of the value of wind gusts (e.g. if the wind is gusting for 6 kts , add 3 kts to the final approach speed). In such conditions I would also recommend to only use 1st stage of flaps for increased manoeuvrability. In very strong winds (20 kts and more), use retracted flaps for the complete approach and roundout.

Parking

Taxi to the apron with flaps retracted (minimum lift). Again, taxi slow for reasons mentioned under "Taxi". Come to a standstill, shut down the engine, insert the parachute rescue system activation handle's safety pin. It is recommended to shut the fuel tank valve.

Conversion tables

kilometres per hour (km/h) - knots (kts) - metres per sec. (m/s)

km/h	kts	m/s	km/h	kts	m/s	km/h	kts	m/s
1,853	1	0,37	63,00	34	18,34	124,16	67	36,15
3,706	2	1,07	64,86	35	18,88	126,01	68	36,69
5,560	3	1,61	66,71	36	19,42	127,87	69	37,23
7,413	4	2,15	68,56	37	19,96	129,72	70	37,77
9,266	5	2,69	70,42	38	20,50	131,57	71	38,31
11,11	6	3,23	72,27	39	21,04	133,43	72	38,86
12,97	7	3,77	74,12	40	21,58	135,28	73	39,39
14,82	8	4,31	75,98	41	22,12	137,13	74	39,93
16,67	9	4,85	77,83	42	22,66	138,99	75	40,47
18,53	10	5,39	79,68	43	23,20	140,84	76	41,01
20,38	11	5,93	81,54	44	23,74	142,69	77	41,54
22,23	12	6,47	83,39	45	24,28	144,55	78	42,08
24,09	13	7,01	85,24	46	24,82	146,40	79	42,62
25,94	14	7,55	87,10	47	25,36	148,25	80	43,16
27,79	15	8,09	88,95	48	25,90	150,10	81	43,70
29,65	16	8,63	90,80	49	26,44	151,96	82	44,24
31,50	17	9,17	92,66	50	26,98	153,81	83	44,78
33,35	18	9,71	94,51	51	27,52	155,66	84	45,32
35,21	19	10,25	96,36	52	28,05	157,52	85	45,86
37,06	20	10,79	98,22	53	28,59	159,37	86	46,40
38,91	21	11,33	100,07	54	29,13	161,22	87	46,94
40,77	22	11,81	101,92	55	29,67	163,08	88	47,48
42,62	23	12,41	103,77	56	30,21	164,93	89	48,02
44,47	24	12,95	105,63	57	30,75	166,78	90	48,56
46,33	25	13,49	107,48	58	31,29	168,64	91	49,10
48,18	26	14,03	109,33	59	31,83	170,49	92	49,64
50,03	27	14,56	111,19	60	32,37	172,34	93	50,18
51,80	28	15,10	113,04	61	32,91	174,20	94	50,72
53,74	29	15,64	114,89	62	33,45	176,05	95	51,26
55,59	30	16,18	116,75	63	33,99	177,90	96	51,80
57,44	31	16,72	118,60	64	34,53	179,76	97	52,34
59,30	32	17,26	120,45	65	35,07	181,61	98	52,88
61,15	33	17,80	122,31	66	35,61	183,46	99	53,42

knots (kts) - metres per second (m/s)

	0	1	2	3	4	5	6	7	8	9
0	0	0,51	1,02	1,54	2,05	2,57	3,08	3,60	4,11	4,63
10	0,51	5,65	6,17	6,66	7,20	7,71	8,23	8,74	9,26	9,77
20	10,28	10,80	11,31	11,83	12,34	12,86	13,37	13,89	14,40	14,91
30	25,43	15,94	16,46	16,97	17,49	18,00	18,52	19,03	19,54	20,06
40	20,57	21,09	21,60	22,12	22,63	23,15	23,66	24,17	24,69	25,20
50	25,72	26,23	26,75	27,26	27,76	28,29	28,80	29,32	29,83	30,35
60	30,86	31,38	31,89	32,41	32,92	33,43	33,95	34,46	34,98	35,49
70	36,00	36,52	37,04	37,55	38,06	38,58	39,09	39,61	40,12	40,64
80	41,15	41,67	42,18	42,69	43,21	43,72	44,24	44,75	45,27	45,78
90	46,30	46,81	47,32	47,84	48,35	48,87	49,38	49,90	50,41	50,90

metres per second (m/s) - feet per minute (100 ft/min)

m/sec.	100 ft/min	m/sec.	100 ft/min	m/sec.	100 ft/min
0,50	1	1,96	10,66	21	41,33
1,01	2	3,93	11,17	22	43,30
1,52	3	5,90	11,68	23	45,27
2,03	4	7,87	12,19	24	47,24
2,54	5	9,84	12,75	25	49,21
3,04	6	11,81	13,20	26	51,18
3,55	7	13,78	13,71	27	53,15
4,06	8	15,74	14,22	28	55,11
4,57	9	17,71	14,73	29	57,08
5,08	10	19,68	15,24	30	59,05
5,58	11	21,65	15,74	31	61,02
6,09	12	23,62	16,25	32	62,92
6,60	13	25,51	16,76	33	64,96
7,11	14	27,55	17,27	34	66,92
7,62	15	29,52	17,78	35	68,89
8,12	16	31,49	18,28	36	70,86
8,63	17	33,46	18,79	37	72,83
9,14	18	35,43	19,30	38	74,80
9,65	19	37,40	19,81	39	76,77
10,16	20	39,37	20,32	40	78,74

ICAN (International Committee for Air Navigation) temperatures, relative pressure, relative density and CAS to TAS correction factors as related to altitude

Altitude		Temperature		Relative pressure	Relative density	Cor. factors
feet	metres	°C	°F			
-2.000	-610	18,96	66,13	1,074	1,059	0,971
-1	-305	16,98	62,56	1,036	1,029	0,985
0	0	15	59	1	1	1
1.000	305	13,01	55,43	0,964	0,971	1,014
2.000	610	11,03	51,86	0,929	0,942	1,029
3.000	914	9,056	48,30	0,896	0,915	1,045
4.000	1219	7,075	44,73	0,863	0,888	1,061
5.000	1524	5,094	41,16	0,832	0,861	1,077
6.000	1829	3,113	37,60	0,801	0,835	1,090
7.000	2134	1,132	34,03	0,771	0,810	1,110
8.000	2438	-0,850	30,47	0,742	0,785	1,128
9.000	2743	-2,831	26,90	0,714	0,761	1,145
10.000	3090	-4,812	23,33	0,687	0,738	1,163
11.000	3353	-6,793	19,77	0,661	0,715	1,182
12.000	3658	-8,774	16,20	0,635	0,693	1,201
13.000	3916	-10,75	12,64	0,611	0,671	1,220
14.000	4267	-12,73	9,074	0,587	0,649	1,240
15.000	4572	-14,71	5,507	0,564	0,629	1,260
16.000	4877	-16,69	1,941	0,541	0,608	1,281
17.000	5182	-18,68	-1,625	0,520	0,589	1,302

air pressure as related to altitude

altitude (m)	pressure (hPa)	pressure (inch Hg)	altitude (m)	pressure (hPa)	pressure (inch Hg)
-1000	1139,3	33,6	1300	866,5	25,6
-950	1132,8	33,5	1350	861,2	25,4
-900	1126,2	33,3	1400	855,9	25,3
-850	1119,7	33,1	1450	850,7	25,1
-800	1113,2	32,9	1500	845,5	25,0
-750	1106,7	32,7	1550	840,3	24,8
-700	1100,3	32,5	1600	835,2	24,7
-650	1093,8	32,3	1650	830	24,5
-600	1087,5	32,1	1700	824,9	24,4
-550	1081,1	31,9	1750	819,9	24,2
-500	1074,3	31,7	1800	814,8	24,1
-450	1068,5	31,6	1850	809,8	23,9
-400	1062,3	31,4	1900	804,8	23,8
-350	1056,0	31,2	1950	799,8	23,6
-300	1049,8	31,0	2000	794,9	23,5
-250	1043,7	30,8	2050	790,0	23,3
-200	1037,5	30,6	2100	785,1	23,2
-150	1031,4	30,5	2150	780,2	23,0
-100	1025,3	30,3	2200	775,3	22,9
-50	1019,3	30,1	2250	770,5	22,8
0	1013,3	29,9	2300	165,7	22,6
50	1007,3	29,7	2350	760,9	22,5
100	1001,3	29,6	2400	756,2	22,3
150	995,4	29,4	2450	751,4	22,2
200	989,4	29,2	2500	746,7	22,1
250	983,6	29,0	2550	742,1	21,9
300	977,7	28,9	2600	737,4	21,8
350	971,9	28,7	2650	732,8	21,6
400	966,1	28,5	2700	728,2	21,5
450	960,3	28,4	2750	723,6	21,4
500	954,6	28,2	2800	719	21,2
550	948,9	28,0	2850	714,5	21,1
600	943,2	27,9	2900	709,9	21,0
650	937,5	27,7	2950	705,5	20,8
700	931,9	27,5	3000	701,0	20,7
750	926,3	27,4	3050	696,5	20,6
800	920,0	27,2	3100	692,1	20,4
850	915,2	27,0	3150	687,7	20,3
900	909,0	26,9	3200	683,3	20,2
950	904,2	26,7	3250	679,0	20,1
1000	898,7	26,5	3300	674,6	19,9
1050	893,3	26,4	3350	670,3	19,8

ICAO standard atmosphere

h (m)	h (ft)	T (°C)	T (°K)	T/T ₀	p (mmHg)	p (kg/m ²)	p/p ₀	r (kgs ² /m ⁴)	g (kg/m ⁴)	d	1/S d	V _s	n*10 ⁶ (m ² /s)
-1000	-3281	21,5	294,5	1,022	854,6	11619	1,124	0,137	1,347	1,099	0,957	344,2	13,4
-900	-2953	20,8	293,8	1,020	844,7	11484	1,111	0,136	1,335	1,089	0,958	343,9	13,5
-800	-2625	20,2	293,2	1,018	835	11351	1,098	0,134	1,322	1,079	0,962	343,5	13,6
-700	-2297	19,5	292,5	1,015	825,3	11220	1,085	0,133	1,310	1,069	0,967	343,1	13,7
-600	-1969	18,9	291,9	1,013	815,7	11090	1,073	0,132	1,297	1,058	0,971	342,7	13,8
-500	-1640	18,2	291,2	1,011	806,2	10960	1,060	0,131	1,285	1,048	0,976	342,4	13,9
400	-1312	17,6	290,6	1,009	796,8	10832	1,048	0,129	1,273	1,039	0,981	342	14,0
300	-984	16,9	289,9	1,006	787,4	10705	1,036	0,128	1,261	1,029	0,985	341,6	14,1
200	-656	16,3	289,3	1,004	779,2	10580	1,024	0,127	1,249	1,019	0,990	341,2	14,3
100	-328	15,6	288,6	1,002	769,1	10455	1,011	0,126	1,237	1,009	0,995	340,9	14,4
0	0	15	288	1	760	10332	1	0,125	1,225	1	1	340,5	14,5
100	328	14,3	287,3	0,997	751,0	10210	0,988	0,123	1,213	0,990	1,004	340,1	14,6
200	656	13,7	286,7	0,995	742,2	10089	0,976	0,122	1,202	0,980	1,009	339,7	14,7
300	984	13,0	286,0	0,993	733,4	9970	0,964	0,121	-1,191	0,971	1,014	339,3	14,8
400	1312	12,4	285,4	0,991	724,6	9852	0,953	0,120	1,179	0,962	1,019	338,9	14,9
500	1640	11,1	284,7	0,988	716,0	9734	0,942	0,119	1,167	0,952	1,024	338,5	15,1
600	1969	11,1	284,1	0,986	707,4	9617	0,930	0,117	1,156	0,943	1,029	338,1	15,2
700	2297	10,4	283,4	0,984	699,0	9503	0,919	0,116	1,145	0,934	1,034	337,8	15,3
800	2625	9,8	282,8	0,981	690,6	9389	0,908	0,115	1,134	0,925	1,039	337,4	15,4
900	2953	9,1	282,1	0,979	682,3	9276	0,897	0,114	1,123	0,916	1,044	337	15,5
1000	3281	8,5	281,5	0,977	674,1	9165	0,887	0,113	1,112	0,907	1,049	336,6	15,7
1100	3609	7,8	280,8	0,975	665,9	9053	0,876	0,112	1,101	0,898	1,055	336,2	15,8
1200	3937	7,2	280,2	0,972	657,9	8944	0,865	0,111	1,090	0,889	1,060	335,8	15,9
1300	4265	6,5	279,5	0,970	649,9	8835	0,855	0,110	1,079	0,880	1,065	335,4	16,0
1400	4593	5,9	278,9	0,968	642,0	8728	0,844	0,109	1,069	0,872	1,070	335	16,2
1500	4921	5,2	278,2	0,966	634,2	8621	0,834	0,107	1,058	0,863	1,076	334,7	16,3
1600	5249	4,6	277,6	0,963	626,4	8516	0,824	0,106	1,048	0,855	1,081	334,3	16,4
1700	5577	3,9	276,9	0,961	618,7	8412	0,814	0,106	1,037	0,846	1,086	333,9	16,6
1800	5905	3,3	276,3	0,959	611,2	8309	0,804	0,104	1,027	0,838	1,092	333,5	16,7
1900	6234	2,6	275,6	0,957	603,7	8207	0,794	0,103	1,017	0,829	1,097	333,1	16,9
2000	6562	2	275	0,954	596,2	8106	0,784	0,102	1,006	0,821	1,103	332,7	17,0
2100	6890	1,3	274,3	0,952	588,8	8005	0,774	0,101	0,996	0,813	1,108	332,3	17,1
2200	7218	0,7	273,7	0,950	581,5	7906	0,765	0,100	0,986	0,805	1,114	331,9	17,3
2300	7546	0,0	273,0	0,948	574,3	7808	0,755	0,099	0,976	0,797	1,120	331,5	17,4
2400	7874	-0,6	272,4	0,945	576,2	7710	0,746	0,098	0,967	0,789	1,125	331,1	17,6
2500	8202	-1,2	271,7	0,943	560,1	7614	0,736	0,097	0,957	0,781	1,131	330,7	17,7
2600	8530	-1,9	271,1	0,941	553,1	7519	0,727	0,096	0,947	0,773	1,137	330,3	17,9
2700	8858	-2,5	270,4	0,939	546,1	7425	0,718	0,095	0,937	0,765	1,143	329,9	18,0
2800	9186	-3,2	269,8	0,936	539,3	7332	0,709	0,094	0,928	0,757	1,149	329,6	18,2
2900	9514	-3,8	269,1	0,934	532,5	7239	0,700	0,093	0,918	0,749	1,154	329,2	18,3



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ALPHA Trainer checklist

Before start-up

Fuel system drain	PERFORMED
Doors	CLOSED
Rudder pedals & headrest position	SET
Seat belts	FASTENED
Parachute rescue system safety pin	REMOVED
Pitot tube protection cover	REMOVED
Brakes	SET
Flaps	Retracted
Battery switch	CHECK ON
Instruments	CHECKED

Engine start-up

Area in front of aircraft	CLEAR
Fuel valve	OPEN
Throttle	IDLE
Choke	AS REQUIRED
Master / battery switch	ON
Avionics / generator switch	ON
Magnetos	ON
Starter	ENGAGE
AC/NAV lights	ON
Airbrakes*	CALIBRATE and CHECK

After start-up

Warm up at	2000 RPM for 2 min, then 2500 RPM
Magneto RPM drop	VERIFIED, MAX 500 RPM
Engine & Propeller check COM, NAV	Check parameters, RPM within limits SET

Before takeoff

Fuel valve	CHECK OPEN
Doors	CHECK CLOSED
Flight controls	CHECKED, free and correct
Flaps	EXTENDED +15°
Fuel (Booster) pump (switch)	ON
Elevator trim	SET NEUTRAL

After takeoff

Elevator trim	SET
Flaps	RETRACTED 0°
Fuel (Booster) pump (switch)	OFF

Descent - Approach

Throttle	SET as required
Fuel (Booster) pump (switch)	ON
Flaps	AS REQUIRED (respect speed limits)
Instruments/ Landing light	SET/ON

Landing

Throttle	IDLE or AS REQUIRED (idle at touch down)
Flaps	SET FOR LANDING AS REQUIRED

Shutdown

Brakes	SET
Flaps	RETRACTED 0°
Airbrakes *	RETRACTED
AC lights	OFF
Fuel (Booster) pump	OFF
Magnetos	OFF
Avionics / generator switch	OFF
Master / battery switch	OFF
Fuel valve	CLOSED

*if applicable

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Warranty statement/voids

For applicable warranty terms and conditions please refer to www.pipistrel-aircraft.com.



Pipistrel d.o.o.
Goriška cesta 50a
SI-5270 Ajdovščina
Slovenija - EU

t +386 (0)5 3663 873
f +386 (0)5 3661 263
info@pipistrel-aircraft.com
www.pipistrel-aircraft.com

Pipistrel Italia
Via Fratelli Rusjan, 26
34070 Savogna d'Isonzo (GO)
Italia - EU

t +39 39 3703207623
info.italia@pipistrel-aircraft.com
www.pipistrel-aircraft.com

Pipistrel Vertical Solutions d.o.o.
Vipavska cesta 2
SI-5270 Ajdovščina
Slovenija - EU

t +386 5 366 38 73
f +386 5 366 12 63
info@pipistrel-aircraft.com
www.pipistrel-aircraft.com